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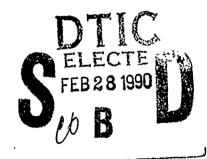
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PEACETIME REPLACEMENT AND CRASH DAMAGE FACTORS FOR ARMY AIRCRAFT

ROBERT L. BENSON
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FINAL REPORT

MAY 1989



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PEACETIME REPLACEMENT

AND CRASH DAMAGE

FACTORS FOR ARMY AIRCRAFT

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May 1989

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Item 19. ABSTRACT (Continued)

Linear regression analysis has been generally successful in generating the PTRF and CD factors for most Army aircraft systems. Cumulative flying hours are regressed against cumulative Class A or Class B accidents. The Class A accident regression generates the PTRF. The Class B accident generates the CD factor. These factors are calculated from the regression equation. The slope of the regression line fitted to Class A accidents and flying hour data points is the PTRF. The slope of the regression line fitted to Class B accidents and flying hour data points is the CD factor.

Non-Linear regression analysis techniques have been applied to those systems that do not fit the linear model well. A typical polynomial equation is:

$$Y = B_0(X)^3 - B_1(X)^2 + B_2(X) - B_3$$

Statistical Analysis System (SAS) software is used to model these systems.



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I. INTRODUCTION.

- A. Army aircraft mishaps are among the highest concerns of Army management. Loss of an aircraft is very expensive in terms of hardware as well as personnel costs. The Army invests significant amounts of money into training aviators. The Worldwide Aviation Logistics Conference (WALC) meets annually at the U.S. Army Aviation Systems Command (USAAVSCOM) in St. Louis, Missouri. One of the issues that the WALC addresses is Safety. The WALC uses factors to assess probable accident rates for proposed flying levels. The Peacetime Replacement Factor (PTRF) relates flying hours to the expected number of aircraft losses. The Crash Damage (CD) Factor relates flying hours to the expected number of crash damaged aircraft. Flying hours are expressed in a per 100,000 basis. Annual updates are made to these factors by the Operational Systems Analysis Division of the Directorate for Systems and Cost Analysis at AVSCOM.
- B. Army aircraft mishaps are defined by AR385-40 with specific dollar thresholds. The Class A accident is the most serious. It involves total loss of an aircraft with associated fatalities and/or injuries. Class A mishaps have a threshold of \$500,000 or greater. Class B accidents are defined as those between \$100,000 and less than \$500,000. II. BACKGROUND. Linear Regression analysis has been used successfully to generate the PTRF and CD factors for most Army aircraft systems. Cumulative flying hours are regressed against cumulative Class A or Class B accidents. The Class A accident regression generates the PTRF. The Class B accident generates the CD factor. These factors are calculated from the regression equation. The slope of the regression line fitted to Class A accidents and flying hour data points is the PTRF. The slope of the regression line fitted to Class B accidents and flying hour data points is the CD factor. AVSCOM Regulation 710-7 requires the Directorate for Systems and Cost Analysis to review and provide updates to the PTRF factors.

III. DATA BASE.

A. DATA SOURCES.

- 1. The Fort Rucker Aviation Safety Center is the primary source for all data in this study. The Safety Center maintains records on Class A, B, C, D and E accidents. They also keep records on injury and damage losses. Class A accidents are the most serious because they often result in injury and loss of life.
- 2. Appendix A contains the data base used in the latest update of the Crash Damage (CD) and Peacetime Replacement Factors (PTRF) provided to the 1989 WALC Conference. As mentioned before, Class A trends drive the PTRF factor and Class B accidents drive the CD factor. The Safety Office at AVSCOM provides this data through an electronic link. The Safety Office provides data on flying hours for rotary and fixed wing aircraft.
- B. MODEL DATA BASE. Data on Class A and B accidents along with associated flying hours are input via Statistical Analysis Systems (SAS) software to the Scientific and Engineering (S&E) computer system at AVSCOM. Data has been collected from 1974 through 1988 and is maintained in a SAS file. The SAS programs used for the linear model is contained in Appendix B. Typically, data is input in a free format manner. Variable names are assigned to the data elements, i.e., Class A, Class B, and flying hours. Variables are input to the appropriate SAS procedure. General Linear Models (GLM) is the procedure used for a linear regression. Non-Linear (NLIN) is the procedure used for fitting a curved line to accident and flying hour data points.

IV. ACCIDENT RATE MODELING.

A. LINEAR REGRESSION METHODS.

- 1. The GLM procedure shown in Appendix B uses a "straight-line" fit to minimize the distance of actual data from the fitted line. It does this through a least squares method. In general, the GLM procedure has been successful in providing an indication of future accident rates. There are many variables which play a part in aviation rafety. Some of these variables are:
 - a. Flying hours.
 - b. Maintenance.
 - c. Training.
 - d. Experience Levels.
 - e. Aircraft Complexity.
- 2. Flying hours are not necessarily the most important variable. However, they are relatively easy to capture for most systems. There has been a satisfactory correlation shown between flying hours and accident rates. Variables such as training are much harder to capture. Aircraft complexity is hard to quantify. Maintenance is somewhat more difficult to capture than flying hours. This is an area that should be investigated further. It may be possible to model accident rates with a multiple regression analysis model. This approach would be a logical next step to further refine the present technique. Currently, flying hours are used as an independent variable. Accident rates, i.e., Class A and Class B are the dependent variables.

B. UH-1 LINEAR RESULTS.

1. Figure 4.1 contains a plot of the UH-1 cumulative Class A accidents versus cumulative flying hours. The general model that generates the straight line fit is plotted with the asterisks. The equation of the linear model is:

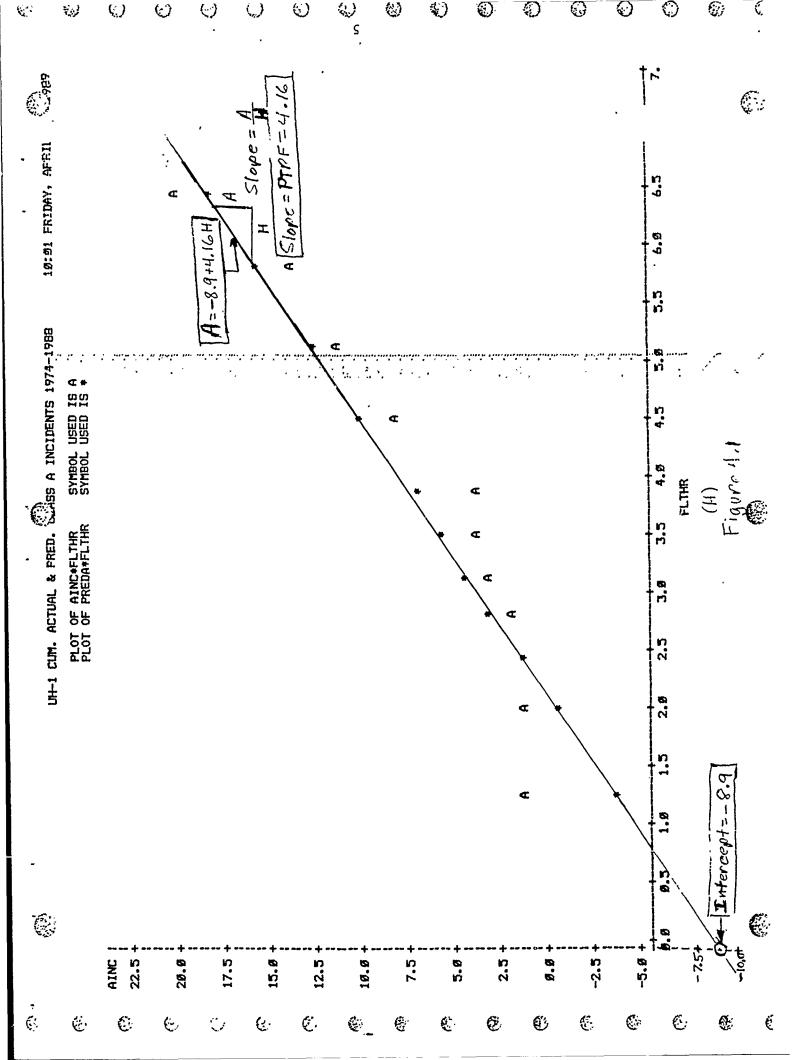
Y = a + bX

where: Y = Cumulative Numbers of Class A or B Accidents (Calculated Value)

b = Slope of the Line (Units of Accidents/Units of flying hours)
This is the Crash Damage factor for Class B accidents and the Peacetime Replacement Factor for Class A accidents

X = Cumulative Number of Flying Hours

2. The SAS output listings for the UH-1 and the other aircraft studied are in Appendix B. The listing for the UH-1 shows that it had an R-squared statistic of .935 for the Peacetime Replacement Fa or and an F Value of 159.23. The R-Square is calculated by dividing the sum of squares for the model by the corrected total. The corrected total is the sum of the model sum of squares and the error sum of squares. Since the calculated value of F exceeds the table value at a 95% confidence level, the linear model's fit is significant at that confidence level.



C. NON-LINEAR RECRESSION METHODS.

- 1. Class A or B accident rate data that did not lend itself to the linear methods was input to a non-linear model. The SAS procedure used for this approach is NLIN. Appendix contains the program listing for those aircraft with data that did not fit the linear model well.
- 2. The following aircraft were modeled with a non-linear regression equation:

OH-6A UH-1 (all models except H)

OH-58 (A, B&C Models) UH-1H

TH-55

3. The NLIN procedure requires the user to input a hypothesized model. The first derivative of each constant in the model must also be specified. A typical model used was a "Cubic" equation. The Cubic equation is so named because the largest exponential term in the model is of degree 3. The general form of the model used in this study was:

 $A = B0(H)^3 - B1(H)^2 + B2(H) - B3$

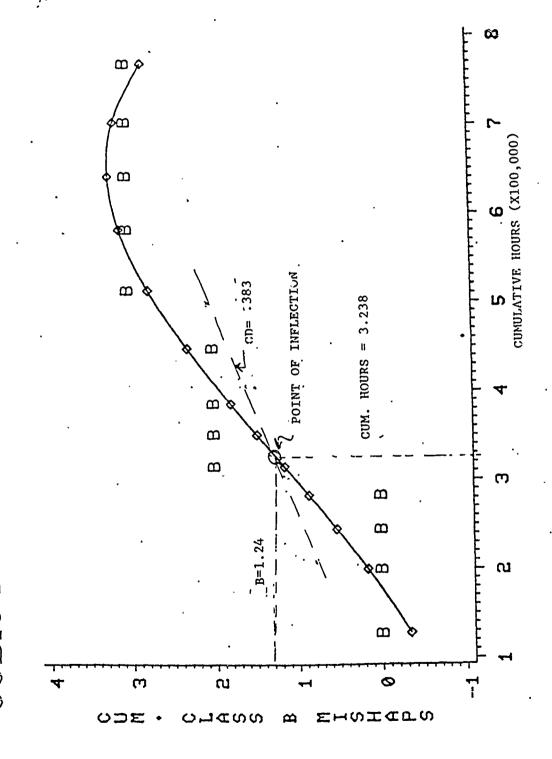
where: BO, B1, B2 & B3 = Constant Coefficients generated by the SA3 Model

A = Number of Cumulative Class A or B Accidents

H = Number of Cumulative Flight Hours

4. Figure 5.1 contains an example of a non-linear regression model applied to the UH-1 Class B Data Base. Non-linear equations are contained in **Table 1** for the UH-1, UH-1H, OH-58, OH-6 and the TH-55 aircraft. These are all cubic equations that have been fit with the SAS NLIN procedure. This type of equation will have a number of minimum and maximum points equal to one less than the power of the equation. For example, a cubic equation will always produce two minimum/maximum points.

CUBIC FIT UH-1 PRED. CLASS B INCIDENTS



D. NON-LINEAR REGRESSION METHODS. Since a non-linear curve fit is not a straight line the slope is not a constant. This makes it impossible to generate a replacement or crash damage factor directly from the regression equation. Therefore, another approach is to use the area of the curve which is the "high" point or maximum for the range of data being studie. This is commonly known as the local maximum.

The first derivative solution of the non-linear equation will give the points where the slope of the curve equals zero. Two of these points are locations on the curve where the minimum and maximum points will occur. The high point of the fitted curve would give a relative maximum area in which to find a point slope factor. The point slope factor will be the slope of a straight line which touches the curve at the point where the curve changes from an increasing to a decreasing slope. This point is also known as the point of inflection. It is found by taking the second derivation of the curve function and setting it equal to zero.

TABLE 1
NON-LINEAR MODELS

AIRCRAFT	MODEL	<u>R</u> ²	
UH 1H1	$B =0279(H)^3 + .27(H)^20295(H)751$.960	23.7
UH-1H	$A =00004(H)^3 + .008(H)^2 + .08375(H) + 4.311$.996	26€.8
OF-6A	$A =1218(H)^3 + .856(H)^2 - 1.332(H) + .449$.931	13.58
OF-582	$A =00013(H)^3 + .0089(H)^21028(H) + .2599$.959	23.36
TH-55	$A = .07974(H)^3 - 1.715(H)^2 + 11.776(H) + 5.619$.9995	2210.6

FOOTNOTES: 1/ Includes all models except UH-1H.

2/ Includes A, B, & C Models.

V. NON-LINEAR MODELING RESULTS.

A. GENERAL. The aircraft systems shown in Table 2 were modeled using the non-linear approach described in Section IV of this report. In general, there was an improvement observed in the fit of the non-linear versus the linear model. Table 2 also shows the R-squared value with the linear and with the non-linear methods. Table 2 also contains the point slope factor that was generated by the non-linear method. The R-squared values are significantly better with the non-linear equation. This is because the line will fit the discrete increases in cumulative incidents much better than a straight line. A straight line fit is much more limited in its capabilities.

TABLE 2 LINEAR/NON-LINEAR MODELING COMPARISON

AIRCRAFT		FACTOR	R-S	QUARED	% Improvement Non-Linear Model
	Linear	Non-Linear	Linear	Non-Linear	Over Linear Model
<u>1</u> /					
UH-1H	.572	.383	.784	.960	22.4
UH-1H	.491	.512	.902	.996	10.4
OH-6	.368	.417	.761	.931	22.3
<u>2</u> / он-58	.072	.460	.884	.959	8.5
TH-55	1.305	4.36	.595	.9995	68.0

FOOTNOTES: 1/ Includes all models except UH-1H.

2/ Includes A, B & C Models.

B. <u>UH-1 RESULTS</u>. The UH-1 aircraft systems analyzed consisted of all UH-1 other than the "H" Model. The "H" model was analyzed on its own because it represents the largest number of aircraft in the UH-1 fleet. Class B accidents were studied for this system. Appendix B-1, page 1, contains the data for the UH-1 system. It was found that a cubic equation, i.e., where the highest variable exponent is equal to three fit well. The derived equation is:

$$B = -.0279(H)^{3} + .271(H)^{2} - .0295(H) - .751$$
 (5.1)

where:

B = Cumulative Number of Class B Accidents

H = Cumulative Flying Hours

The equation (5.1) increases from the y axis up to the point where cumulative flying hours equal to 323,800. At this point on the curve, cumulative accidents are equal to 1.32. The slope of the curve is then equal to the number of accidents (A) divided by the number of hours (H).

Slope =
$$\frac{A}{H(per 100,000)}$$
 (5.2)
= $\frac{1.32}{3.238}$
= .407

The second derivative of the regression equation is taken and set equal to zero. The resulting value of H (Hours Flown) is at the point where the model equation changes from an increasing to a decreasing slope. The following process is used:

(1) Take the first derivative of equation 5.1 with respect to H. $\frac{dB_{-}}{dH} = -.0837H^{2} + .542H + .0295 (5.1.1)$

(2) Take the second derivative of equation 5.1.1 with respect to H.

$$\frac{d^2B}{dH^2} = -.1674H + .542 \quad (5.1.2)$$

(3) Set equation 5.1.2 equal to zero and solve for the value of H.

(4) Calculate the value of B from equation 5.1 when H equals 3.238.

$$B = -.0279(3.238)^3 + .271(3.238)^2 + .0295(3.238) - .75$$

= -.9472+2.8413+.0955-.75
= 1.2397

(5) Calculate the value of the slope at the point where B equals 1.2397 and H equals 3.238.

Slope =
$$\frac{1.2397}{3.238}$$
 = .383

Therefore, the new peacetime replacement factor for the UH-1 becomes .383. Figure 4.1 contains the linear plot for the UH-1. As a comparison, Figure 5.1 contains the plot of the UH-1 non-linear curve. The CD factor of .383 is shown also.

C. UH-1H RESULTS.

1. The UH-1H aircraft system was modeled with a cubic equation.

Class B accidents were studied to determine a better fit than was possible with the linear model previously used. It was found that the following equation fit very well, resulting in a R-Squared value of .9995:

$$B = -.00004(H)^{3} + .0:81(H)^{2} + .08376H + 4.311 (5.3)$$

- 2. The following procedure is used to calculate the CD factor:
 - a. Take the first derivative of equation 5.3 with respect to H.

$$\frac{dB}{dH} = -.00012(H)^2 + 0612(H) + .08376 (5.3.1)$$

b. Take the second derivative of equation 5.3 with respect to H.

$$\frac{d^2B}{dH} = -.00024(H) + .0162 (5.3.2)$$

c. Set equation 5.3.2 equal to zero and solve for H.

$$-.00024(H)+.0162 = 0$$

 $.0162 = .00024(H)$
 $H = 67.5$

d. Calculate the value of B in equation 5.3 when H equals 67.5.

$$B = -.00012(67.5)^{3} + .0081(67.5)^{2} + .08376(67.5) + 4.311$$
$$= 34.57$$

e. Calculate the slope of a straight line at the point where H equals 67.5 and B equals 34.57. This will become the new CD factor.

$$CD = \frac{34.57}{67.5} = .512$$

The calculated value of F for the model used in equation 5.3 is *4.7.

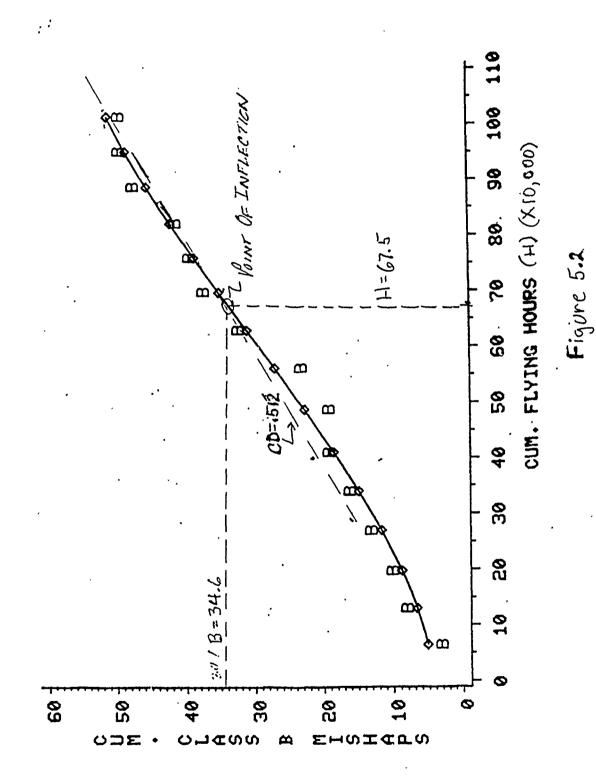
The table value of F at a 95% confidence level is:

$$F(4,11,.95) = 3.36$$

Since the calculated value of F exceeds the table value, it can be inferred that the regression model exhibits a significant level of fit at 95% confidence.

Figure 5.2 contains the plot of the non-linear curve generated by equation 5.3 and the crash damage factor location on the curve.

CUBIC FIT UH-1H CUM. CLASS B INCIDENTS



D. OH-58 RESULTS.

1. The OH-58 systems analyzed consisted of the "A" and "B" models. The Class B accident data was input to SAS with a cubic equation model. The R-square statistic was calculated at .959 for this aircraft. The derived equation from the SAS program was:

$$B = -.00013H^3 + .0089H^2 - .1028H + .2599$$
 (5.4)

- 2. Calculation of Crash Damage (CD) Factor: The point of inflection for equation 5.4 is calculated by setting the second derivative equal to zero and solving for the value of H (cumulative hours).
 - a. First Derivative with respect to H:

$$\frac{dB}{dH} = -.00039(H)^2 + .0178(H) - .1028$$

b. Set Second Derivative equal to zero and solve:

c. Calculate A when H = 22.8 hours from equation 5.3

$$B = -.00013(22.8)^{3} + .0089 (22.8)^{2} - .1028(22.8) + .2599$$

$$B = -1.54 + 4.667 - 2.34 + .2599$$

$$B = 1.05$$

d. Slope = $\frac{\text{Cumulative Accidents}}{\text{Cumulative Hours (Per 100,000)}}$

$$=\frac{1.05}{2.28}$$

$$= 0.46$$

e. CD factor for the OH-58 equals 0.46.

3. Several points can be calculated with the slope factor of .46:

Flight Hours (Cumulative, Per 100,000)	Class B
1.0	.46
1.5	.69
2.0	.92
2.5	1.15
3.0	1.38

4. Figure 5.3 shows the cumulative hour/accident curve, the point of inflection corresponding to a slope of .46 where Accidents equal 1.05 and hours equal 280,000. Also the straight line generated by a slope of .46 has been plotted for the data pairs shown above.

E. OH-6 RESULTS.

1. The OH-6 Class B accidents and flying hours were modeled by the following cubic equation:

$$A = -.1218H^3 + .8564H^2 - 1.1332H + .449 (5.5)$$

The first and second derivatives of equation 5.5 were calculated to find the point of reflection.

$$\frac{dA}{dH} = -.3654H^2 + 1.712H - 1.1332$$

$$\frac{d^2A}{dH} = -.7308H + 1.712$$

2. Setting the second derivative equal to zero and solving for H (Cumulative Hours) results in the point of reflection on the curve shown in Figure 5.6.

$$-.7308H+1.712 = 0$$

H = 2.343

3. The value of H equal to 2.343 red in equation 5.6 to calculate the number of accidents for this flying revel.

$$A = -.1218(2.343)^{3} + .856(2.343)^{2} - 1.1332(2.343) + .449$$

$$A = -1.567 + 4.699 - 2.655 + .449$$

$$A = .926$$



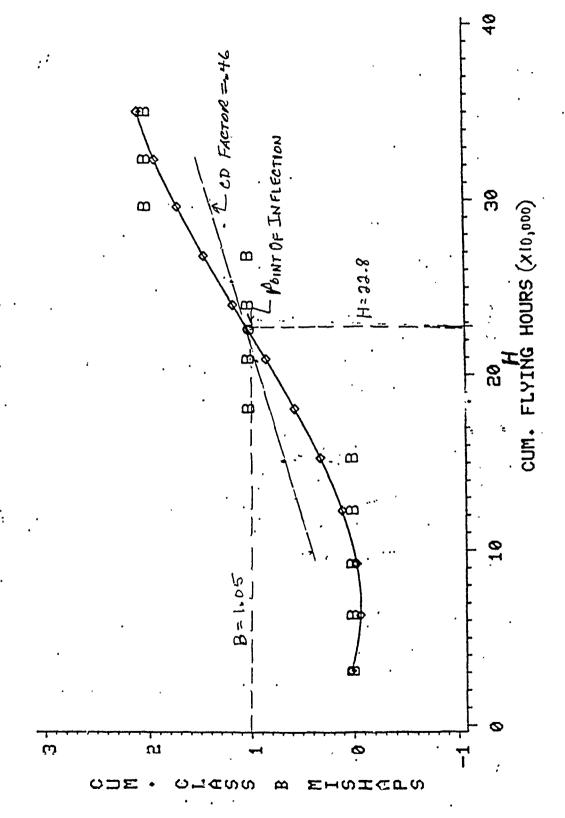


Figure 5.3

4. The slope of a straight line at the point on the curve where H=2.343 and A=.976 becomes the Crash Damage (CD) factor for the OH-6.

$$CD = \frac{.976}{2.343} = .4166$$

- 5. Several points on the straight line with a slop equal to .4166 were calculated as in the case of the OH-58 system. This line is plotted on Figure 5.4.
- 6. The R-Squared statistic for the curve generated by the model equation (5.4) is equal to .931. This is a big improvement over the linear model fit of .761. The F test will determine if the curve fitted by the non-linear model is significant. If the calculated F statistic is greater than the table value with a level of significance equal to .93 and 4 degrees of freedom. The table value equals

$$F(4,9,.95) = 3.63$$

7. The calculated value of F from the regression model is:

$$F = \frac{Sum \text{ of Squares (Regression)}}{Sum \text{ of Squares (Residual)}}$$

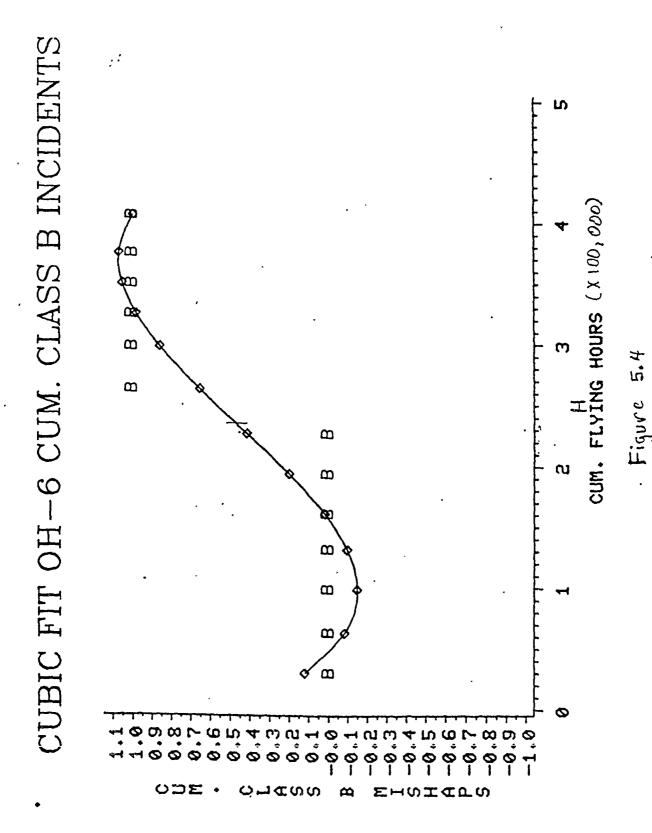
$$= \frac{5.5885}{.4115} = 13.58$$

8. Since the calculated value of F is greater than the table value, the regression model exhibits a significant level of fit at the 95% level of confidence.

F. TH-55 RESULTS.

1. The TH-55 model A aircraft was modeled for the occurrence of Class A accidents. A cubic equation was used to model this system:

$$A = .0797H^3 - 1.715H^2 + 11.776H + 5.619$$
 (5.8)



- 2. The first and second derivative of equation 5.8 were calculated to find the point of inflection.
 - a. First Derivative of H With Respect to A

$$\frac{dA}{dH} = .2391H^2 - 3.43H + 11.776$$

b. Second Derivative of H With Respect to A

$$\frac{d^2A}{dH} = .4782H - 3.43$$

3. The next step is to set the second derivative of equation 5.6 equal to zero and solve for H.

$$.472H - 3.43 = 0$$

 $H = 7.17$

4. The value of H equal to 7.17 is used in equation 5.8 to calculate the number of accidents (A).

$$A = .0797(7.17)^3 - 1.715(7.17)^2 + 11.776(7.17) + 5.619$$

 $A = 29.3776 - 88.1663 + 84.4339 + 5.619$
 $A = 31.264$

Figure 5.6 displays the regression plot for the TH-55 non-linear fit.

5. The slope of a straight line at the point where H equals 7.17 and A equals 31.264 becomes the Peacetime Replacement Factor for the TH-55A.

$$PTRF = \frac{31.264}{7.17} = 4.36$$

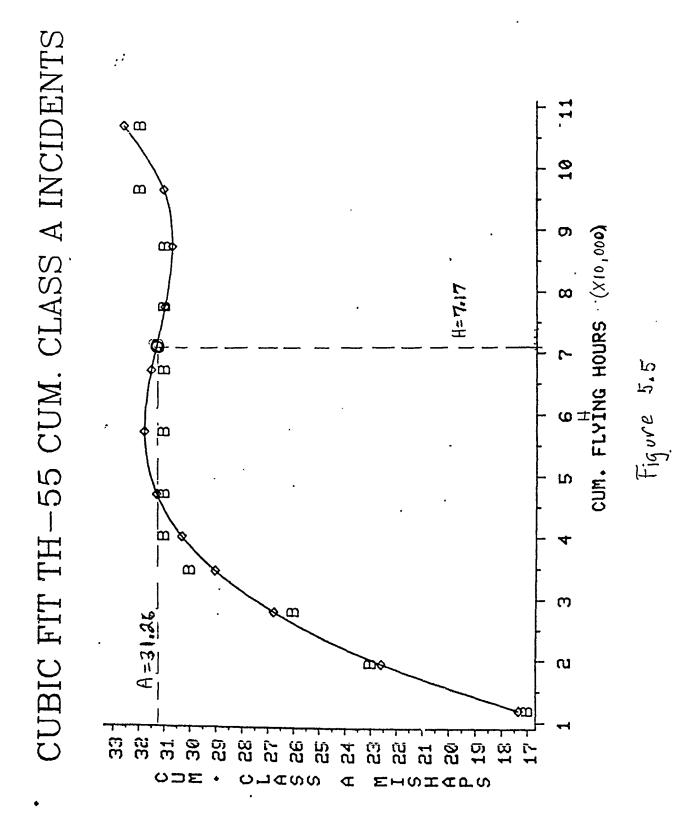
6. The R-Squared statistic was also improved for the system. It increased from .595 to .9995. The table value of F at a 95% confidence level is:

$$F(4,8,.95) = 3.84$$

7. The calculated value of F is:

$$F = 10,203.38/4.6156 = 2,210.6$$

- 8. Since the calculated value of F far exceeds the table value, it can be inferred that the regression model exhibits a significant level of fit at 95% confidence.
- 9. Figure 5.5 contains the non-linear curve fitted to TH-55 Class A mishaps.



VI. CONCLUSIONS.

- A. <u>DATA BASE</u>. The least squares regression analysis tool has proven to be a useful method for assessing Class A and B accident rates. The linear method is usually acceptable when there is sufficient data history present. The systems with a smaller occurrence of accidents do not generally fit this type of model well. Examples of these are the Class B accidents for the UH-1, OH-58, OH-6 and TH-55 aircraft. For these aircraft, Class B occurrences are few and far between. Several years may pass before one occurs. Currently, most systems in the data base have fifteen years of data available. This is enough data to get a pretty good idea of trends in accident rates.
- B. LINEAR MODEL. The General Linear Models procedure available with Statistical Analysis System (SAS) does a good job of describing accident rates for the aircraft studied. SAS provides enough statistics for the user to judge whether the model fit is satisfying. SAS will also provide a 95% confidence level for each point on the regression line.

C. NON-LINEAR MODEL.

1. The Non-Linear (NLIN) Procedure available with SAS provided a lot of flexibility for the user. It will allow the user to choose almost any type of hypothesized model to fit his or her data. Some models which may be tried are:

$$a. Y = aX^b$$

b.
$$Y = ae^{bX}$$

c.
$$Y = b_0 X^n + b_1 X^{n-1} + ... + b_n X^{n-(n+1)}$$

2. Along with the SAS plotting procedures, the NLIN procedure provides a fast way to try any equation desired and see the results on a terminal instantly.

VII. CAUTIONS. None of these tools (GLM and NLIN SAS procedures) are meant to provide infallible answers to the user. As with any automated tool, the output is only as good as the input. High R-squared values do not tell the complete story about what is going on with a data base. They should be balanced with other statistical tests of significance. Ideally, the residuals from the regression equation should be plotted and analyzed. Also, the first equation that fits should not be immediately accepted as the best possible model. Several others should be tried and the results compared. Also, one should not extrapolate beyond the range of the data base because the results obtained may be unrealistic.

REFERENCES

- 1. AR 385-40, Accident Reporting and Records, 1 April 1987, w/AMC Supplement 1, 10 August 1987.
- 2. AVSCOM Regulation 710-7, Worldwide Aviation Logistics Conference (WALC), 13 March 1989.

APPENDIX A

ACCIDENT AND FLYING HOUR

DATA BASE

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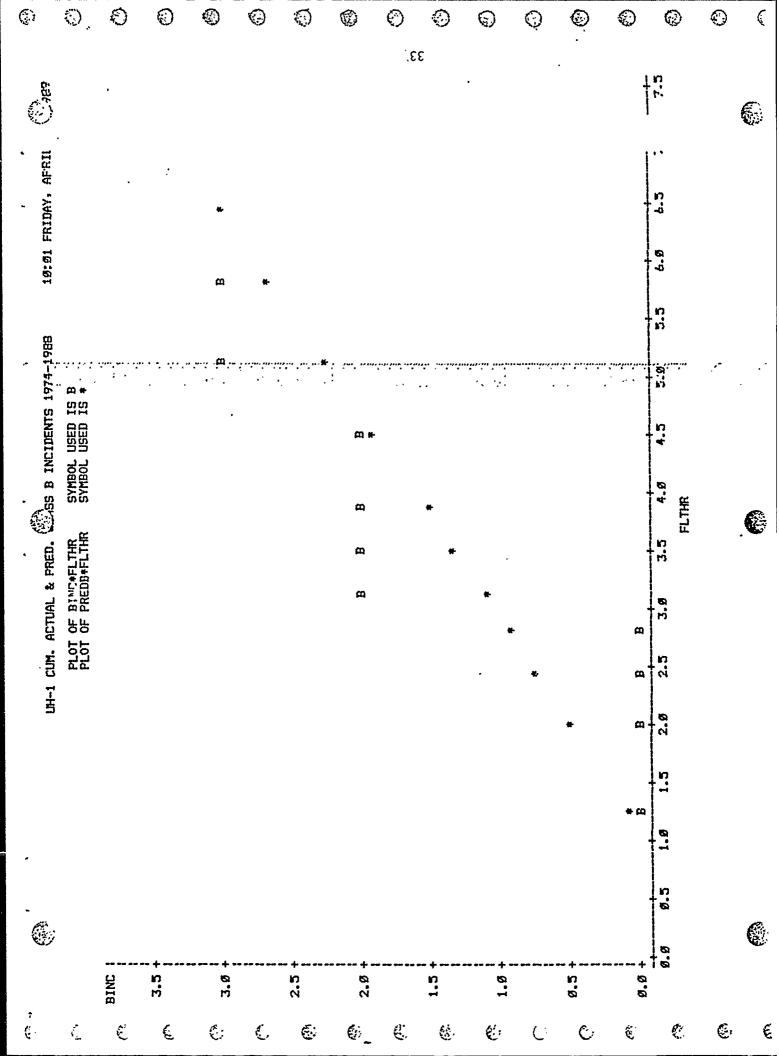
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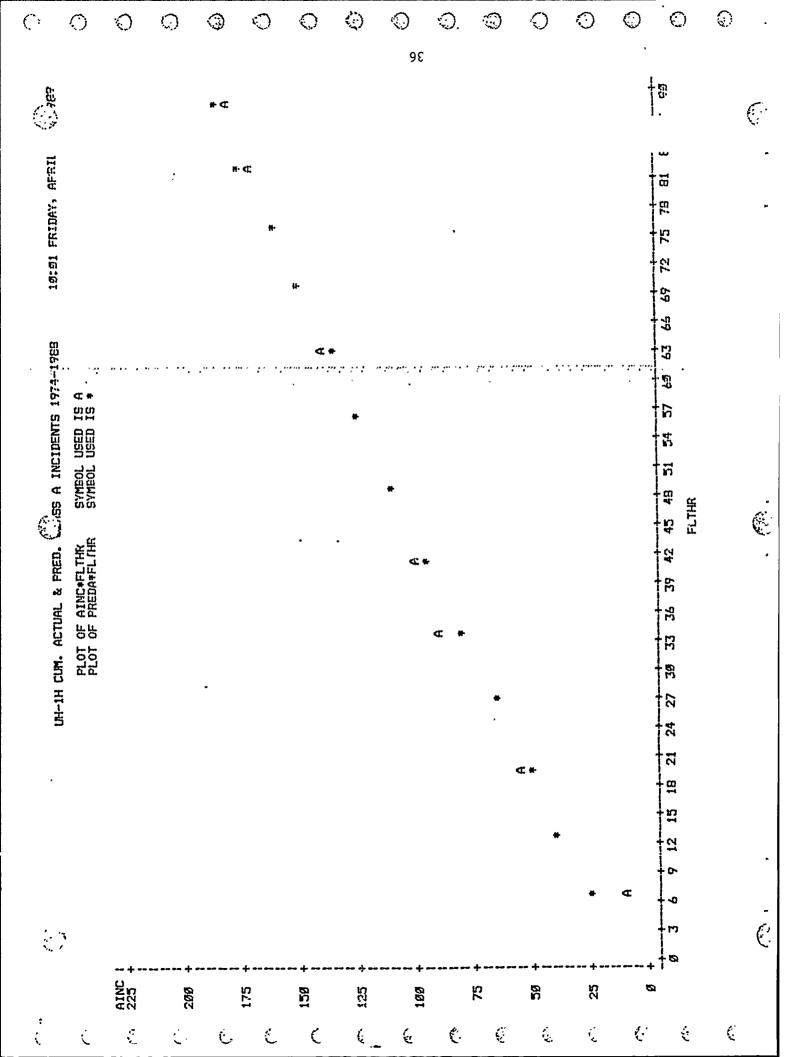
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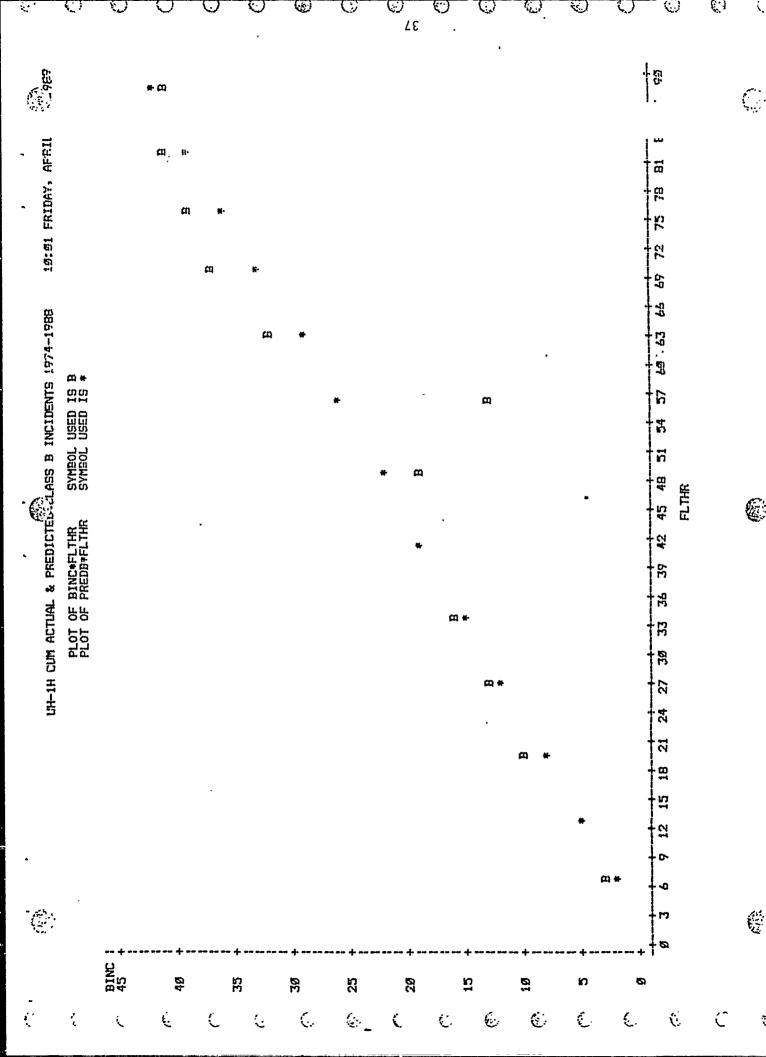
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GENERAL LINEAR MODELS PROCEDURE

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HENEBOL I TNEOR			MEAN	1674,38248501	10,39		F VALUE	161.03	PR > 1Ti	e. 6195 6.6601	Œ	เร็นใน เรียน	 		4.04 UNA	######################################		
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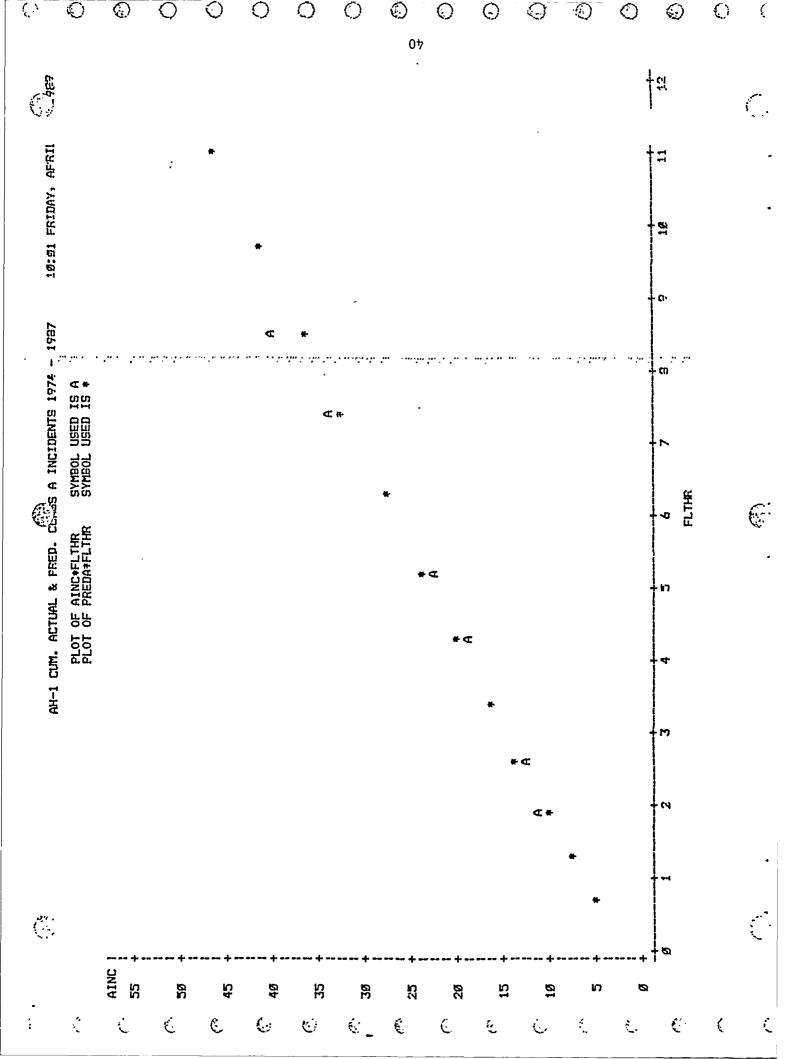
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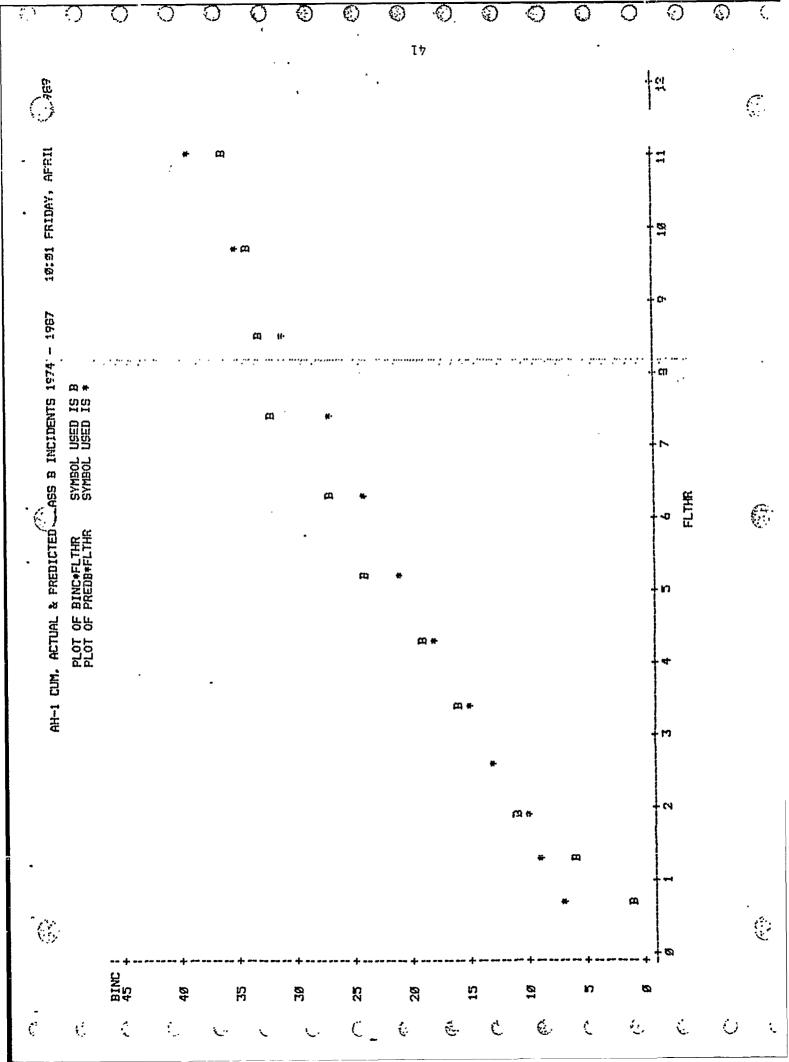
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E			GEN	GENERAL LINEAR MODELS	DELS PROCEDURE	a		•		
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10:91 FRI			F .	9. 9651	ROOT MEE	1.35429674	TYPE III 5S	927.55812829			UPPER 75% (FOR MEAN			15,85943949 17,65423143 26,15391969 22,5525559	
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GH-47	FRAL LINEAR		MEGN	907.00	1.83		F VALUE	494.61	PR > ITI	g. 8956 g. 8661	Œ				เรีย
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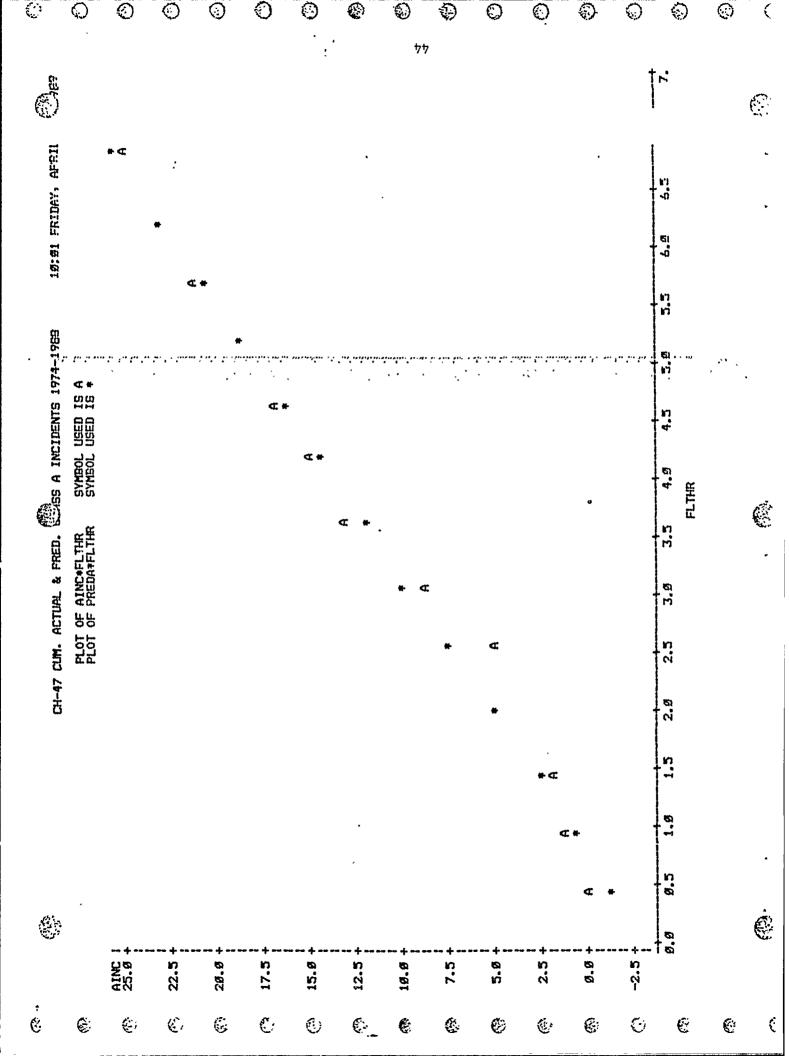
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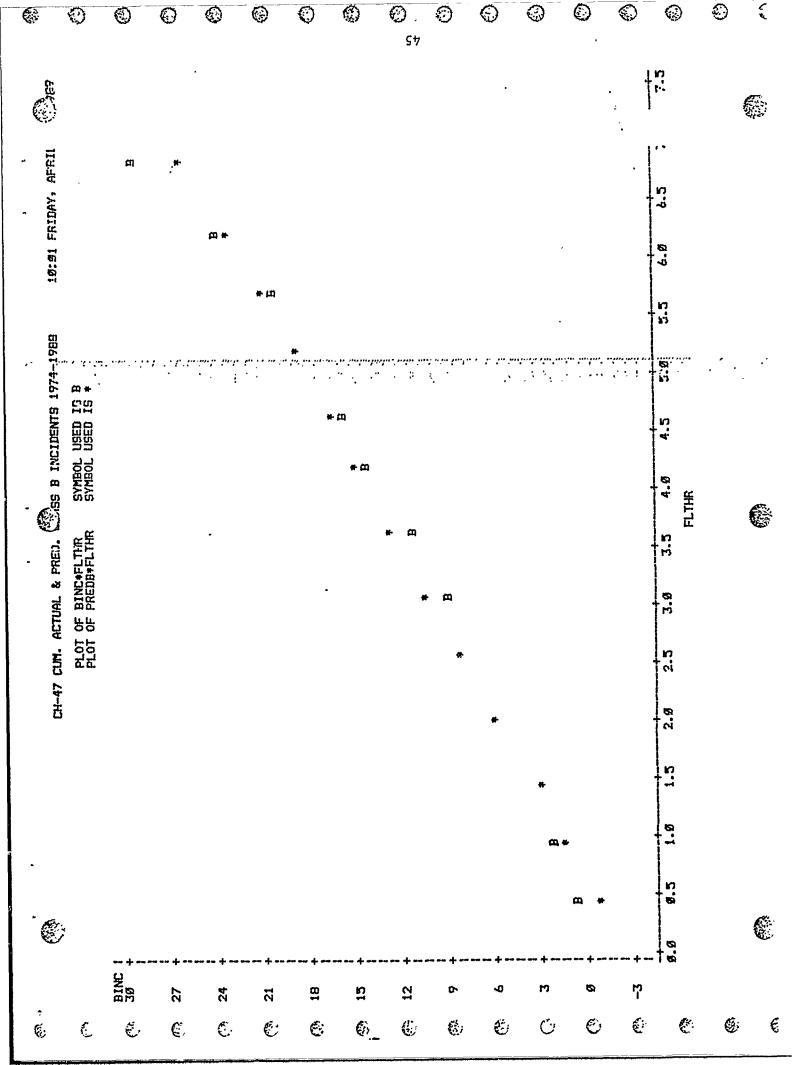
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*CH-54 LY.ZAR FIT	MODELS FRO		MEAN SQUARE	14,63677567	0,20784557		PR > F	Ø. 9Ø31	118	લાલા	RESIDUAL	-9.14296269 -9.4948595 -9.1759867 -9.454518889 -9.55624893 -9.55624893 -9.34542817 -9.34342817 -9.3434817 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125 -9.55638125
40-HD*	GENERAL LINEAR MODELS FROCEDURE		MEAN	14,63	0.20		F VALUE	78.42	PR > iTi	8.5443 8.8881	.	డిడి.ఇడి.డి.ఇ.ఇ.ఇ.ఇ.డి.డి. - ఇ.జి.జి.జి.జి.జి.జి.జి.జి.జి.జి.జి.జి.జి.
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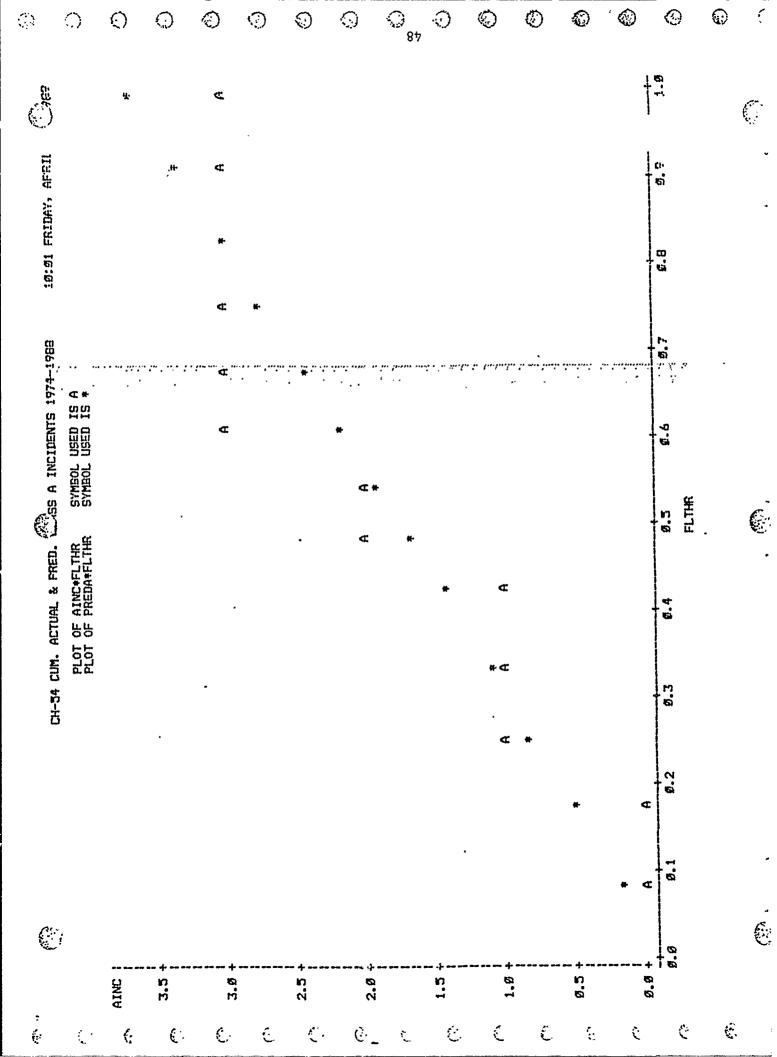
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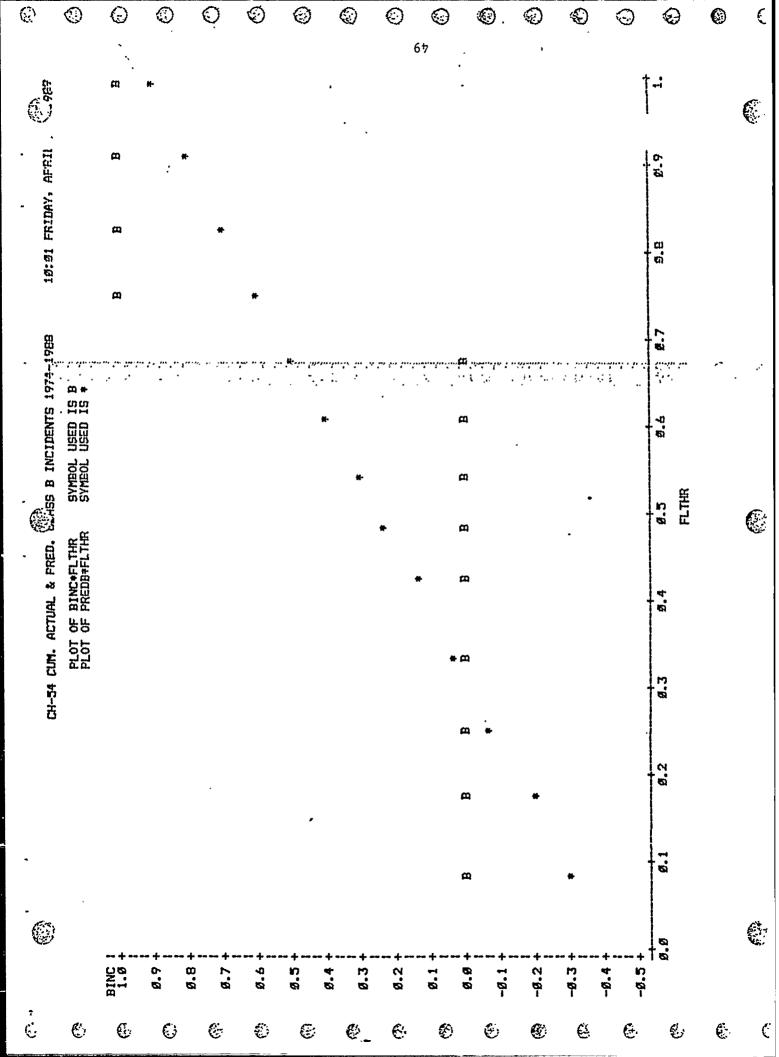
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iðiði FRIDAY,			PR > F	8. <u>9691</u>	ROOT MEE	.56648991	TYPE III SS	21690265.95			LIPPER 95% CL FOR MEAN	5.92945543 5.73925371 6.43364698	7.8191796 7.8191796 8.5152981 9.3175852	18.1782369 11.9413245 11.7293371	13.7972998		
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04-6 LICAR FIT	LINEAR MODELS PROCEDURE		SQUARE	99,39386912	Ø,3289898\$		PR >	8. 9691	ots eg	ଷଣା	RESIDUAL.	-6.35242961 -6.14469733 -6.97849212	1327765 7487798 1882148 18279Ø5	537415 537415 599813 787107	1885792 1411499	-6.6998888 3.53999788 -6.6998349 4.85983281 -6.13842543	1241829
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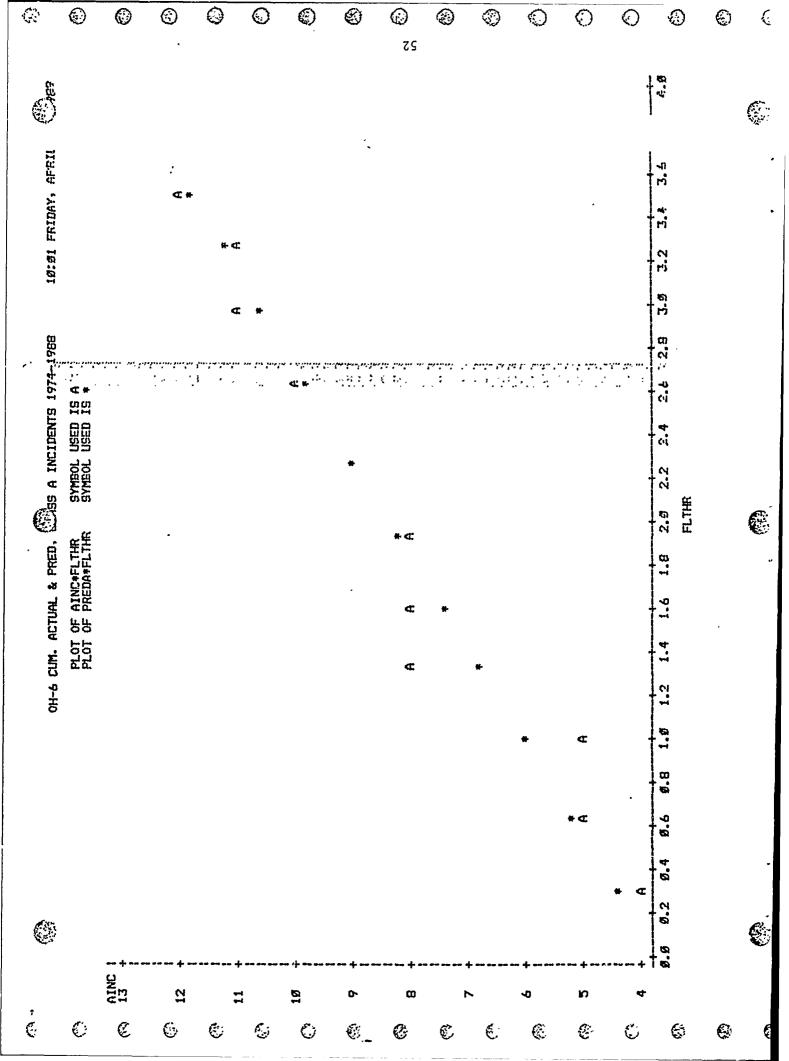
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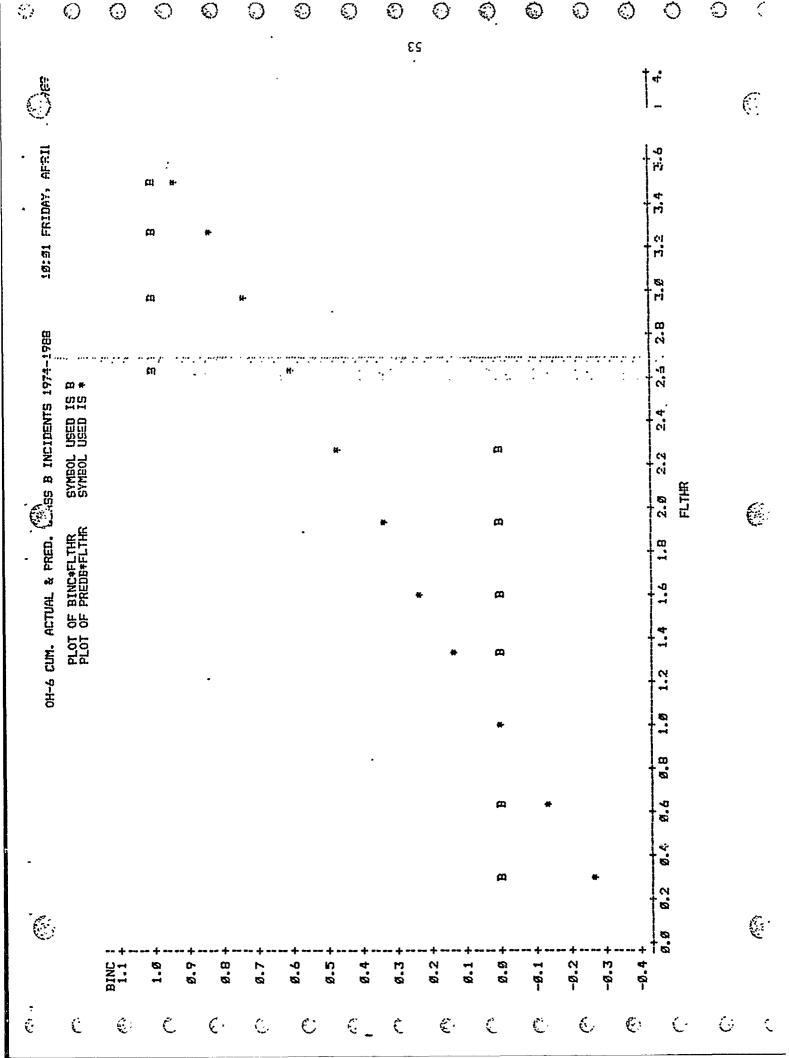
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	A:INC	4 '		11	12	쑴	•• •	ESTIMATE	-7.54£14237 3.438£8747	OBSERVED VALUE	୍ ଫେଉଟେଉଟେଉଟ 15. ଡେଉଟେଉପରେଉଟ 27. ଜୟକ୍ଷରକ୍ଷର	13.000000000000000000000000000000000000	61. 600000000 74. 600000000 82. 60000000 98. 6000000	105.00000000 117.00000000 123.00000000	SUM OF RESIDUALS SUM OF SOURTED RESIDUALS SUM OF SOURTED RESIDUALS — I FRESS STATISTIC FIRST ORDER AUTOCORRELATION
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<i>;</i>	JEDURE :	*	F VALUE	53, 58	Ne k	,	山	•••	STD ERROR OF ESTIMATE	g.18486773 . §.69737256	LOWER STA	-6.71418646 -9.43631171 -9.18631475 8.8633475	6.3125 6.5315 6.7315	1.1355			v
*OH-58 LALAR FIT	MODELS PROCEDURE		MEAN SQUARE	7,89953537	6.09404914		PR > F	0. <u>9</u> 0 <u>61</u>	TS.	Giùi	RESIDUAL	6.33459988 6.12383597 -6.85828954 -6.3646716	2297 8 75 751 4 59 <u>8</u> 7326255	5025115 5213459 5598215 513889	5643569 5966789	-6.00000000 1.03454055 -6.00000000 1.41989.:3	3527371 3327491
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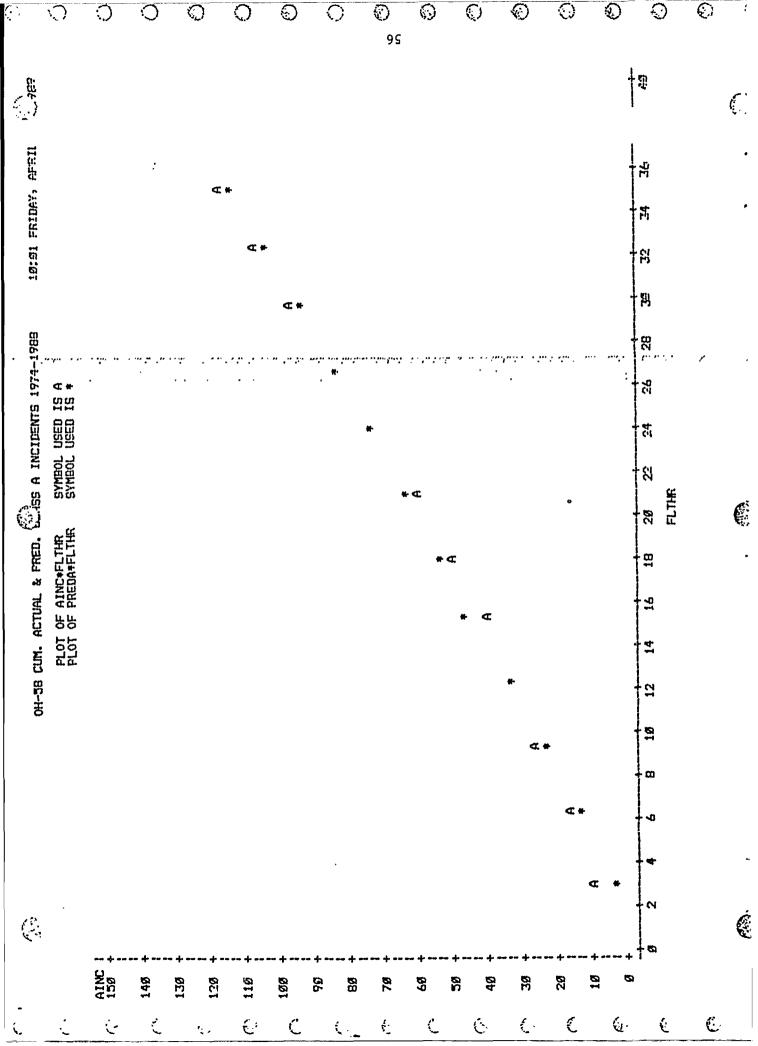
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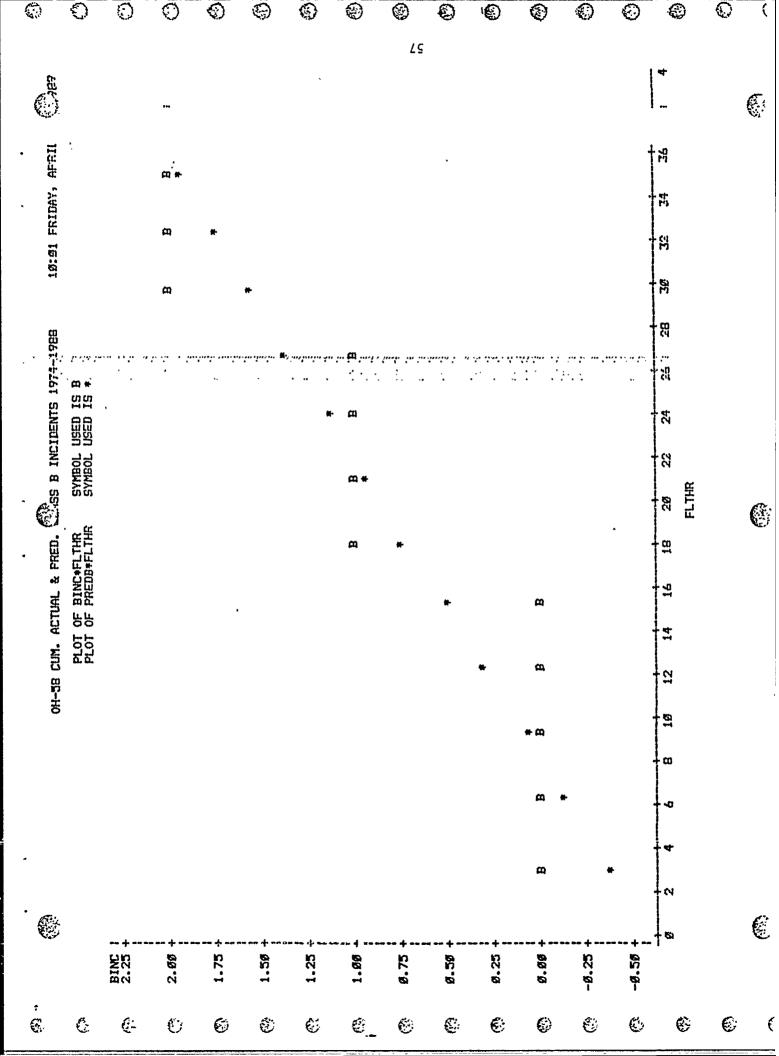
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OAR FIT	MODELS PROC		SQUARE	7,0699	14400		PR > F	Ø, 8833	STS	୯୮ସ	RESIDUAL	195299 129193	92738 182437 18388 13388	12904 164496 132169	326677	141998 141998 172973 181898 182149
*TH-53 LEGAR FIT	ERAL LINEAR		MEAN SC	178,3945690	12,16954400		F VALUE	14.67	PR > IT!	8.8881 9.8833	PES	-5.29	1.64992738 -5.29782437 5.98795879 4.9913388	2.2.5. 2.2.5. 2.2.5. 2.2.5.	. 15. 15. 15. 15. 15. 15. 15. 15. 15. 15	121.60543998 121.60543998 2.00060609 177.32972975 2.07083099 1.57282149
	SE)		SUM OF SOUMRES	178,39456662	121.60543998	366. 6666666	TYPE I SS	178.39456992	T FOR HØ: PARAMETER=Ø	79.67	PREDICTED VALUE	22,29595299	24,35997262 25,20782437 25,91294121 26,80886612	29,41248996 38,75735384 32,83781153 33,23832169	34, 38326477	- ERROR SS ION
		E AINC	H	-	50	11	Ä	#	ESTIMATE	28.74288448 1.38577192	OBSERVED VALUE	17. 6086808 6	26. 99989999 21. 89889999 31. 88899999 31. 98899999	31. 959989998 31. 959889989 31. 95889989 31. 95889998	3:2. 89988988	SUM OF RESIDUALS SUM OF SOLARED RESIDUALS SUM OF SOLARED RESIDUALS - EF PRESS STATISTIC FIRST ORDER AUTOCORRELATION DURBIN-MATISON D
<i>©</i>		DEPENDENT VARIABLE:	SOURCE	MODEL	ERROR	CORRECTED TOTAL	SOURCE	FLTHR	PARAMETER	INTERCEPT FLTHR	OBSERVATION	~ (เมษมข	~ a o - 2 = 5		PRESS STREET OF PRESS STREET O
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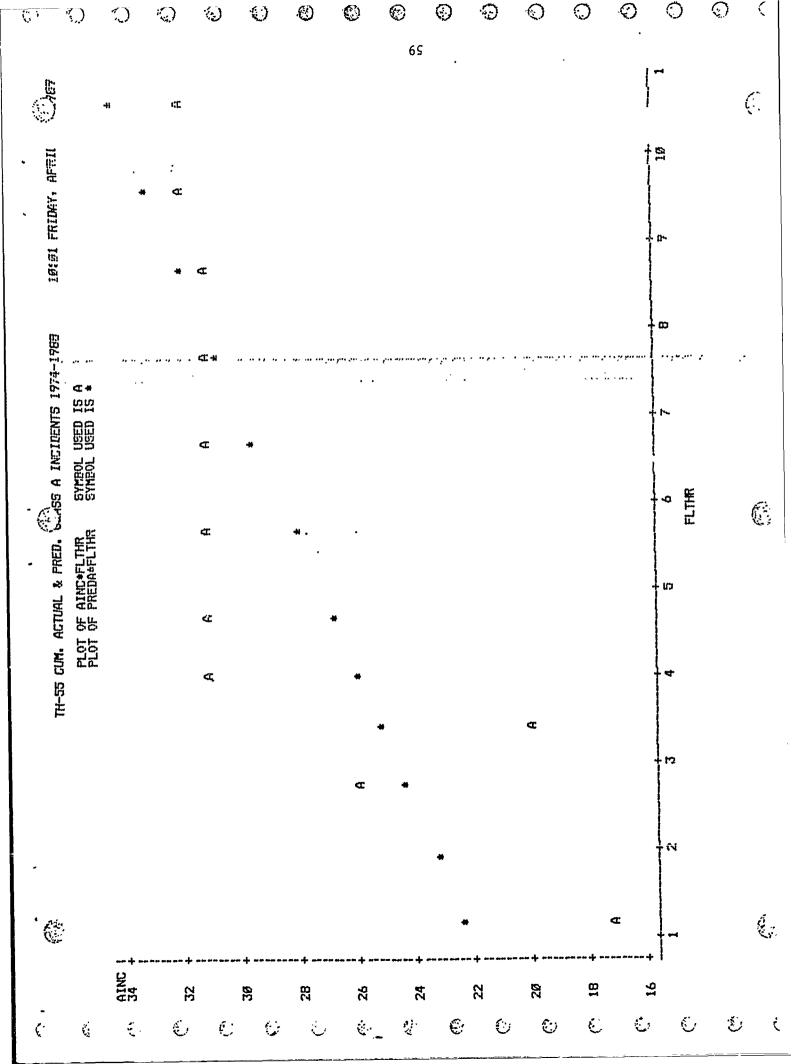
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E	avaes		32,78944531	4,58428647	1647	•	EDOT MEE		:	FILE
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•	*UH-60 LYZAR FIT	GENERAL LINEAR MODELS PROCEDURE
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€.	ERROR	7	7.34344838	1.04996495	95		ROOT	OT MSE	:	N III	5 ,]
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€ },	OBSERVATION	OBSERVED VALUE	PREDICTED VALIE	RESIDUAL	UAL	LOWER 952 C FOR PEAN	, d	UPPER 95% (FOR MEAN	ᆸᇫ		
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	· ca c	7.000000000000000000000000000000000000	7.69486117 10.86449500	-Ø-69486 -Ø-86449	117 598	4 2564. 9 89649	1992 1387	8.831291 12.63859	55 55 55 55 55 55 55 55 55 55 55 55 55		
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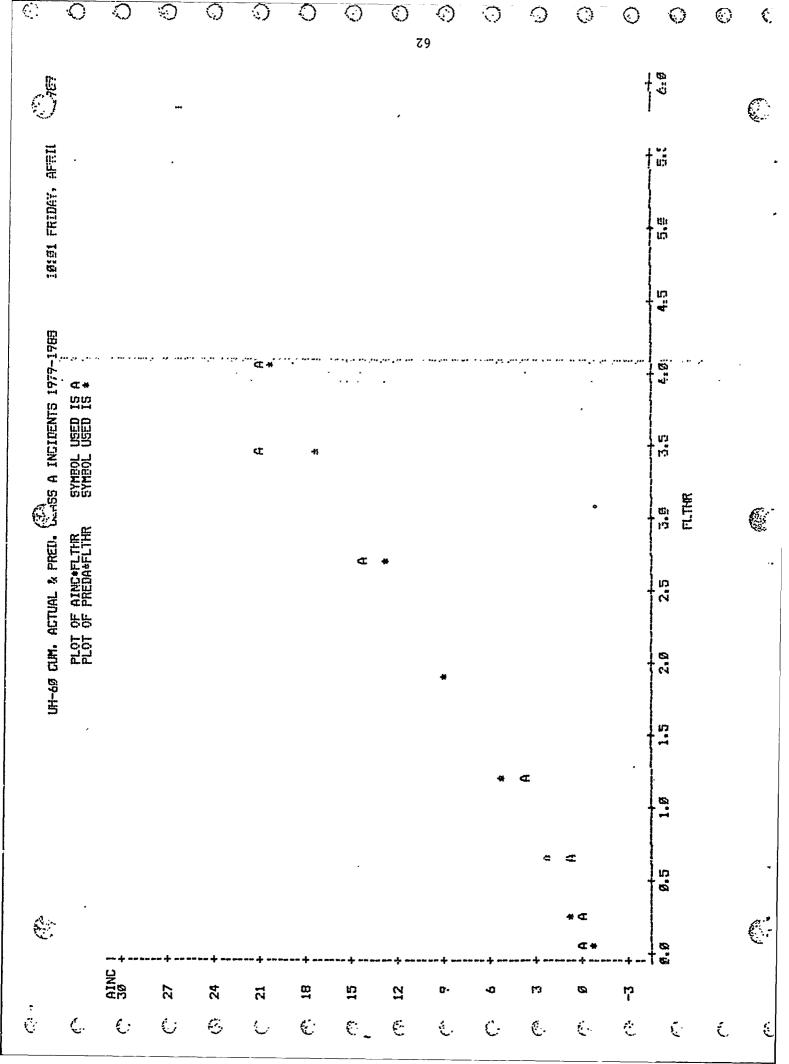
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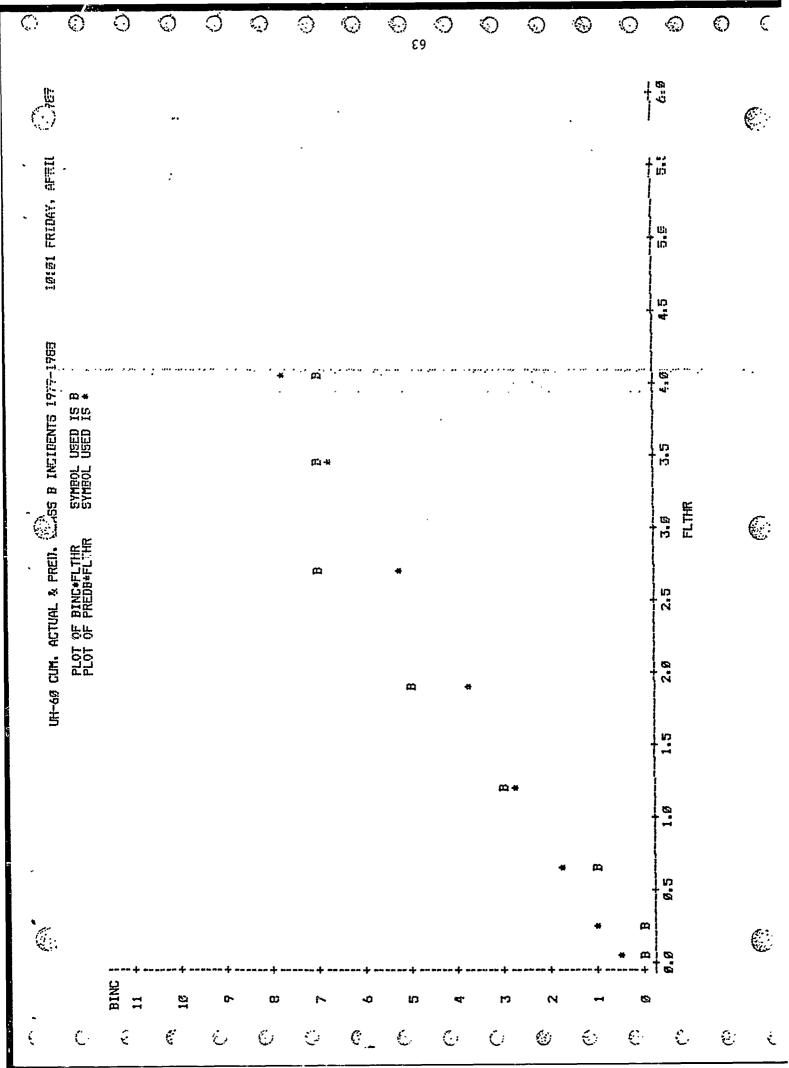
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1988 C-12 CLASS B ACCIDENT CARSH DAMAGE FACTOR REFORT

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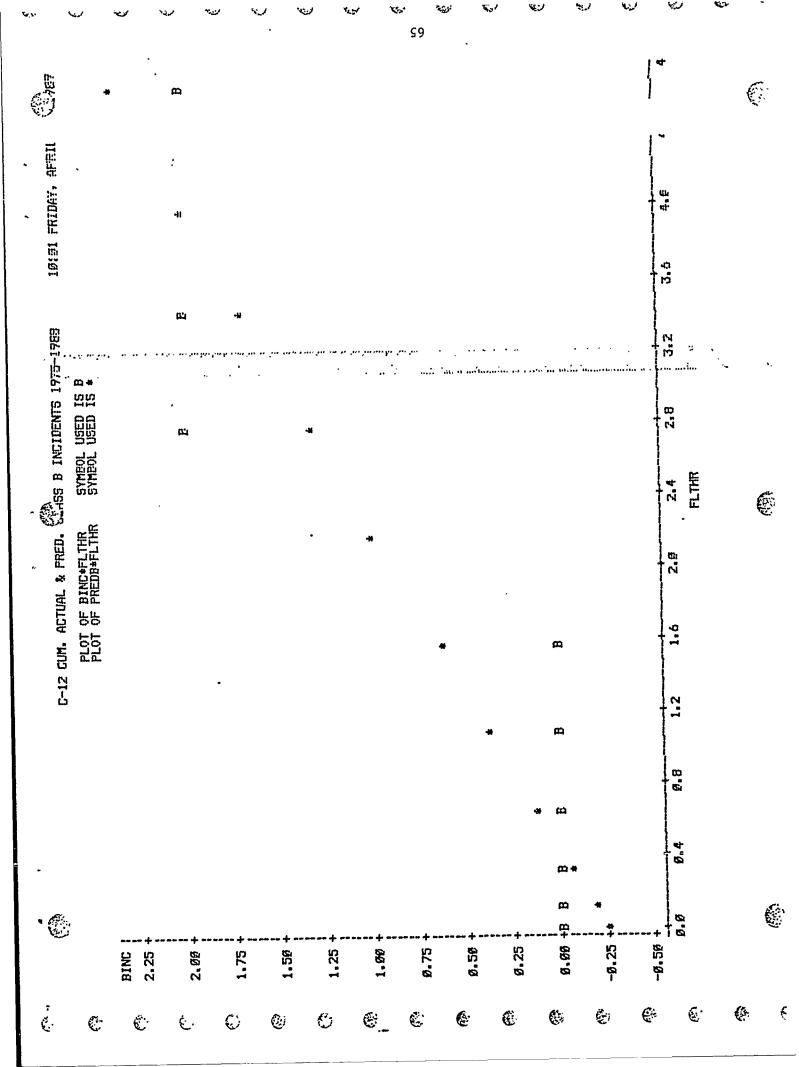
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DEPENDENT VARIABLE: AINC SOURCE SUM OF SQUARES
ERROR 16 4.59113557
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T FOR HØ: PARAMETER ESTIMATE PARAMETER=Ø
INTERCEPT -6.45768329 -1.88 FLTHR 2.55269318 9.37
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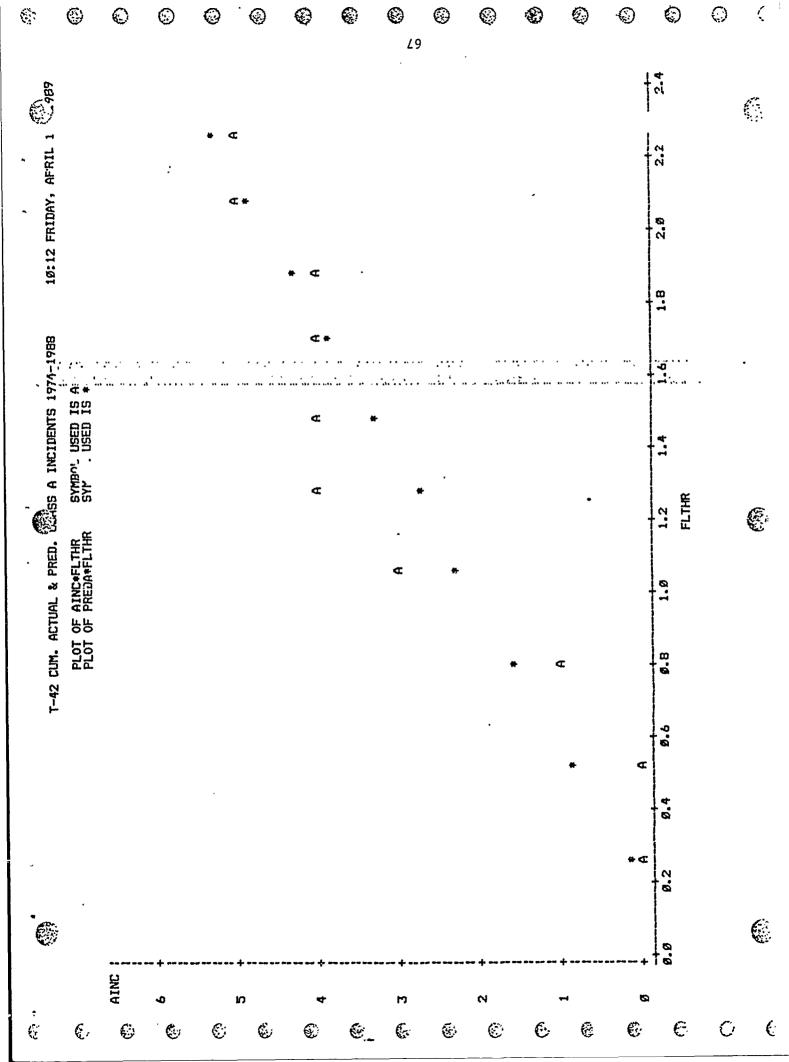
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r, Afril 1			R-SQUPARE	6.984615	;		F VALUE	194.32				10 10 1			0101		
10:12 FRIDAY, AFRIL			PR > F R	B. BBB1 \$	ROOT MSE	88498	TYFE III SS	40.91641364			UPPER 95% CL. FOR MEAN	3.519Ø1735 4.3834Ø491 5.2ØØ838ØØ	5.99147877 6.66939131 7.97263933 7.38772685	7.71251881 8.81495361 8.35485869 8.63554299	8.91993782 8.91993782		
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REPORT	ROCEDURE		F VALUE	164.32			L L	 1	STD ERROR O ESTIMATE	Ø. 5883429973	LOWER FOR		น พูพูจุ	7,77	8.7		,
PTRF 1988 CACTOR REPORT	LINEAR MODELS PROCEDURE		MEAN SOUMRE	46.91641564	0.39221396		PR >	8. 8881			RESIDUAL	.69496238 .67979847 .34474839	.43786974 72488952 .31275478 .61118931	-9.29053616 -9.36475932 0.13359525 -6.11326564	36999648	-\$. \$68688888 4.31433339 \$.\$88688988 6.348 €918 \$.58341832	.87456287
U-8 PTRF 1º	GENERAL LINE		MEA	40.	. 63		F VALUE	104.32	PR > 1T!	8.8966 8.8881						<i>&</i> .4.2.0,20	esi
	75		SUM OF SQUARES	46.91641364	4.31435359	45.23076923	TYPE I SS	40,91641564	T FOR HO: PARANETER=0	3,34	PREDICTED VALUE	2.60406238 3.67079047 4.65525161	5.56293826 6.27591848 6.68724522 6.98889869	7.29#33416 7.56#75932 7.8664#48# 8.1132#364	8.36ØØ648 8.36ØØØ648	S - ERROR SS TION	
		E: AINC	FQ.		11	12	JO	•••	ESTIMATE	1.66896141	OBSERVED	2. ଉଷ୍ଟେଶ୍ୟର 3. ଉଷ୍ଟେଶ୍ଟେଶ୍	7. ชชชชชชชชช 7. ชชชชชชชช 7. ชชชชชชชช 7. ชชชชชชชช	7.00000000 7.00000000 8.00000000 8.00000000	B. ଉଉପସେଥରେ B. ଉଉପସେଥରେ	SUM OF INESTITUALS SUM OF SQUARED RESIDUALS SUM OF SQUARED RESIDUALS - I PRESS STATISTIC FIRST ORDER AUTOCORRELATION	-WATSON D
	٠	DEPENDENT VARIABLE:	SOURCE	MODEL	ERROR	CORRECTED TOTAL	SOURCE	FLTHR	PARAMETER	INTERCEPT FLTHR	OBSERVATION	+db	410.0V		122	SUR OF	DURBIN
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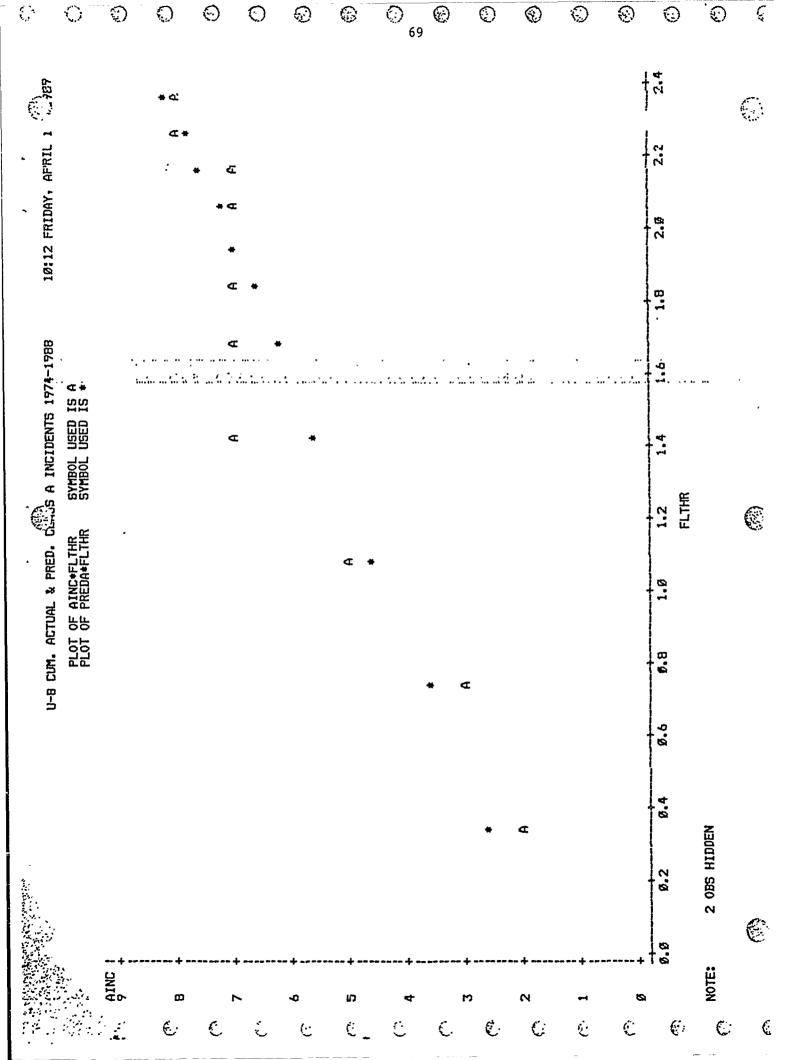
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TEST ACTOR CEST	מייות בויס			ᄣ		ш	7	55 111	59. <i>0</i> 7671267			UPPER 95% CL FOR MEAN	1.21448383 1.72125249	2.6875698 3.18632454 3.67005698	4.2169892E 4.786689Ø3 5.43939Ø46	6.08881589 6.75175981 7.38615545 8.12670449			
•	4			~ &:	0.0001	ROOT MSE	0.59132857	TYPE III	39. 8				संस	7425	MP#	118616 118616			
.:		12 m	, 11 411	<u>.</u>	93		,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,,	· · · · · · · · · · · · · · · · · · ·			598:	ER 95% CL OR MEAN	-0.14068974 Ø.53015174	. 1314388 . 7648253 366566	4537724 9456789 6255316	5.16464121 5.69#23238 6.17834212 6.73437772	**************************************	*** *** ***	٠.
	REPORT	OCEDURE		F VALUE:	168					STD ERROR (ESTIMATE	Ø.344Ø2869 Ø.Ø8466598	LOWER	គ្នា ទ	<i>C</i>	11044	พหงจ		•	
	PTRF 19852/ACTOR REPORT	LINEAR MODELS PROCEDURE		SQUARE	71267	5694B		PR > F	6.0001	ហ		RESIDUAL	5937 64 576212	519758 355469	527163 586914 381646 244188	-0.62672855 -0.22899618 0.21775122 -6.43154111	-6.66666666 3.84636426 -9.68666866	828376 828876 621166	
	PTRF 1983			MEAN SC	59,07671267	Ø.3496694B		VALUE	168.95	:Ti	9.9823 g.gagi	REC	- 53 123			6 4 6 6	2000 P	2.28 3.28 3.18 3.18 3.18 3.18 3.18 3.18 3.18 3.1	
	N-21	GENERAL		ES	29	26	92	SS	29	9		CTED	3704 0212	4985 9758 4531	7817 3986 8334 4499	2855 9616 4878 4111			
				SUM OF SQUARES	59.07671267	3.84636426	62.92387692	TYPE 1	59.07671267	T FOR HØ: PARAMETER=Ø	-6.62 13.88	PREDICTED VALUE	0.5369	2.2261 2.2261 2.7764	3,2936 3,8549 4,4161	5.62672855 6.22899618 6.78224878 7.43154111		7	
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			AINC	띰	4	11	12	TO.		ESTIMATE	-Ø.00780821 1.10049546	OBSERVED VALUE	6. 8888888 1. 88888888	1. 6666666 2. 6696666 4. 669666	4. 6666666 4. 6666666 5. 666666	5. 000000000 6. 0000000000 7. 000000000	SUM OF RESIDUALS SUM OF SQUARED RESIDUALS SUM OF SQUARED RESIDUALS - ER	TISTIC ER AUTOCO TSON D	
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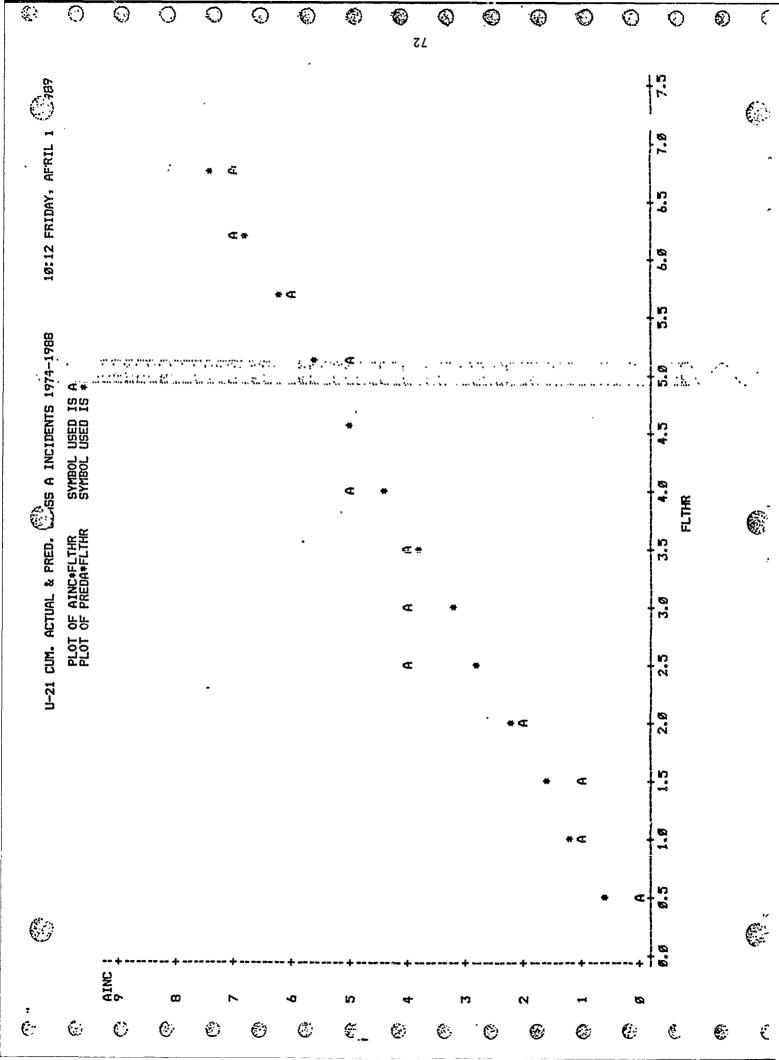
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, 4Y, AFRIL 1			R-SQUARE	£, 8265:76	<i>:</i>		F VALUE	52,43			占	p-024	787 4		125		
, 10:12 FRIDAY,			PR > F	6.0001	ROOT MSE	0.53271427	TYPE III SS	14.87837ø62			UPPER 95% (FOR MEAN	6.9118691 1.1326692 1.3447922	1,5648697 1,7938199 2,8232184 2,291168	2,58124288 2,92333286 3,27125272 3,63135183	3,9789864		
· 1.1	** / ***** *	.i. 1.	·	· · in "		1 53	: · <u>i</u> r ·	· .		· · · · · ·	95% CL MEAN	997ø334 59634ø9 9978888	3279889 5531881 556846 1844195	1.91358322 2.19614545 2.43868508 2.43868508	7681527 344824		, i
PORT	CEDURE		F VALUE	52,43			DF		STD ERROR OF ESTIMATE	Ø.38992751 Ø.87627363	LOWER FOR	<i>8</i> 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	20111 101111		42.2		
CD 1989 ACTOR REPORT	LINEAR MODELS PROCEDURE		MEAN SQUARE	14.87837962	ø.28378449		PR > F	0.0001			RESIDUAL	38868293 59615168 12778929	35157827 57543124 31586856 31419876	-6.24746365 -6.55673876 -6.85496896	56513915 2392951 <i>Ø</i>	-\$.80000000 3.12162938 -\$.00000000 4.31363844 \$.53798523	37672335
U-21 CD 191	GENERAL LINEA		MEAN	14,8	0.2		F VALUE	52.43	PR > IT!	8.9314 8.8881	u -	80.0	8888	6 6 6	666	60040	ш. В
			SUM OF SQUARES	14.87837062	3,12162938	18. 6966666	TYPE I SS	14.87837062	T FOR HØ: PARAMETER=Ø	5.09	PREDICTED VALUE	.3668293 .59615168 .87229671	14842973 42456876 68413944 94580124	2.24746305 2.55673876 2.85496890	43486085 76070499	R SS	
			SUM OF	14.8	3.1	18.8	7	14.8	T F PARA			<i>@</i>	144 444 444	1000r	ממנ	NLS NLS - ERROR ATION	
•		ភ	늄		=======================================	12	片		ESTIMATE	Ø. 82738529 Ø. 55227885	OBSERVED VALUE	5565555 5556666 6566666	99999999999999999999999999999999999999	2. 6666666 2. 66666666 2. 66666666	99999999999999999999999999999999999999	F RESIDUALS F SQUARED RESIDUALS F SQUARED RESIDUALS STATISTIC ORDER AUTOCORRELATION	C)
		ABLE: BINC				ل.				ଭର			તાં ભંજો છે. -	เล่งได้เ	444	SUM OF RESIDUA SUM OF SQUARED SUM OF SQUARED PRESS STATISTI FIRST ORDER AU	BIN-WATSO
` <i>©</i>		DEPENDENT VARIABLE:	SOURCE	MODEL	ERROR	CORRECTED TOTAL	SOURCE	FLTHR	PARAMETER	INTERCEPT FLTHR	OBSERVATION	HOM	ነ ቀ የነሳ	~86~ 8 ;	1212		מחם
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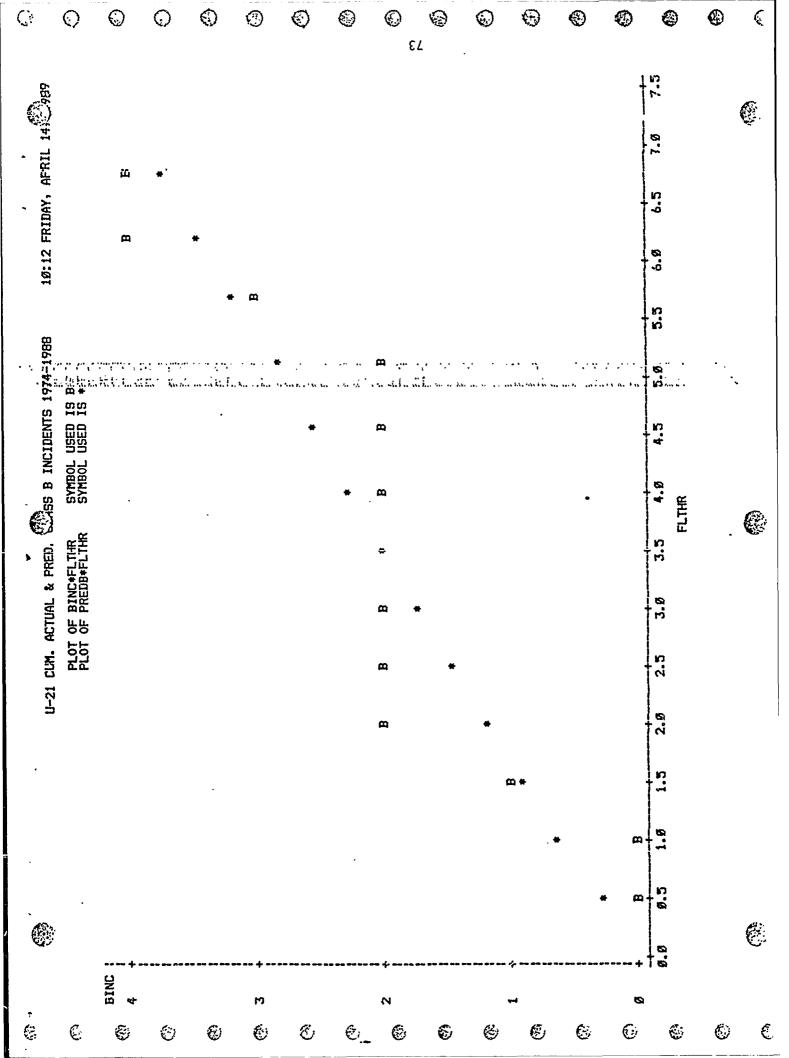
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AFRIL 14			R-SQUP.RE	Ø. 9822:29	<i>:</i>	Ħ	F VALUE	68.88				
19:12 FRIDAY, AFRIL			PR > F R-5	g.ggg1 \$.9	ROOT MSE	ø.85262845	TYPE III SS	442.0032719B			UPPER 95% CL FOR MEAN	1.297ø5996 2.75ø86479 4.33298259 6.02906432 7.59934439 9.29643373 10.94853373 112.48268369 113.71627858 113.71627858 113.766972858 18.766972858
				• ••	,	8	<i>.</i> .	,`	 .	* **	95% CL. MEAN	72569882 96285412 89452646 69919486 42987374 42987374 38693431 83756978 83756978 91814446 91814446
EPORT	CEDURE	,	F VALUE	638.00:		9 702 ,482	 E	· "	D ERROR OF	Ø. 51639817 Ø. 32189588	LOWER 9	-0.72560882 2.80452686 2.80452686 6.429873749 9.9047374 11.38593431 17.9156078 17.9161446 17.91614446
ACTOR RI	fiodels Pro		QUARE	27198	8.72697527		PR > F	g, ggg1	ST2		RESIDUAL	1.71427843 -6.174314853 -1.31468367 -1.31468967 -1.31468967 -6.2346861 -6.5346861 -6.5366861 -6.5366861 -6.5366861 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.536686 -6.53668 -6.53668 -6.53668 -6.5368
OV-1 PTRF 1982 ACTOR REPORT	CENERAL LINEAR MODELS PROCEDURE		MEAN SQUARE	442,00327198	9.726		F VALUE	88.88	PR > IT!	0.6269 0.6661	22	าตอ่าเปลอดตอดตอดตอดตอด นายนายการ
ó	GENE		SUM OF SQUARES	442.00327198	7. '9672802	450.00000000	TYPE I SS	442.00327198	T FOR HØ: PARAMETER=Ø	-2.55 24.66	PREDICTED VALUE	8.28572757 1.85685945 3.57875433 5.36412919 7.81468987 18.7266881 11.92419387 15.82895847 15.82895847 15.8255826 17.88555826 17.88555826 17.88555826 17.88555826 17.88555826
		: AINC	占	+4	137	12	υF	Ħ	ESTIMATE	-1.31714831	OBSERVED VALUE	2. 95388888 2. 98088888 3. 98088888 4. 980888888 10. 98088888 12. 98088888 14. 98088888 14. 98088888 16. 98088888 16. 98088888 16. 98088888 18. 980888888 18. 9808888888 18. 980888888 18. 980888888 18. 980888888 18. 980888888 18. 98088888 18. 98088888 18. 9808888 18. 9808888 18. 980888 18. 9808888 18. 98088 18. 9808 18. 9808
		DEPENDENT VARIALLES	SOURCE	MODEL	ERROR	CORRECTED TOTAL	SOURCE	FLTHR	PARAMETER	INTERCEPT FLT:18	OBSERV TION	SUM OF PRESS

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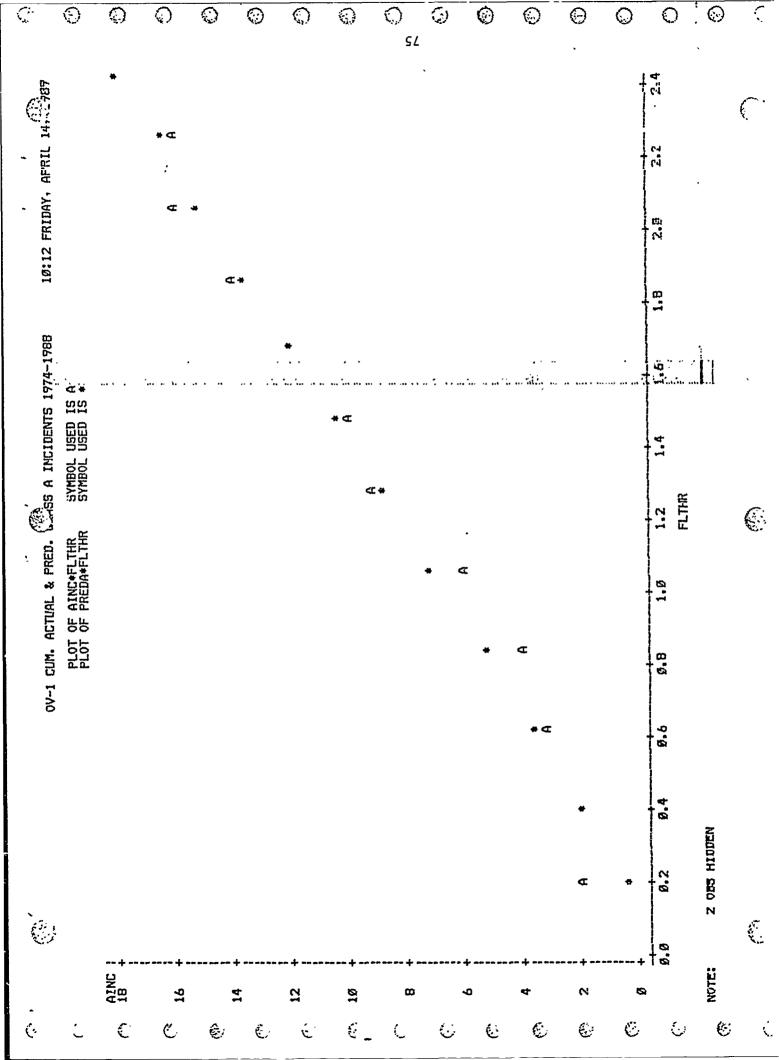
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APPENDIX C

NON-LINEAR SAS OUTPUT LISTINGS

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TOF:
× 4.
       SAS PROGRAM TO FIT A CUBIC EQUATION TO CLASS B INCIDENTS
       FOR THE UH-1 AIRCRAFT. 1989 LINEAR FIT WAS ONLY .7837 *.
DATA UH1REG:
 INPUT YEAR CHOURS HOURS AINC BINC:
   CHOURS=CUM FLIGHT HOURS (X10000);*/
/*
/* AINC = CUM CLASS A INCIDENTS; */
/* BINC = CUM CLASS B INCIDENTS
CARDS:
1977 1.28 1.28 1 9
1978 2.00 .72 1 0
1979 2.45 .45 1 0
1980 2.83 .38 2 0
1981 3.15 .32 3 2
1982 3.51 .36 3 2
1983 3.86 .35 4 2
198~ 4.48 .62 8 2
1985 5.13 .65 11 3
1986 5.82 .69 14 3
1987 6.42 .60 20 3
1988 7.03 .61 23 3
1989 7.69 .66 24 3
                                                             MORE...
PROC NLIN:
     PARMS B0=0 B1=0 B2=0 B3=0;
           MODEL BINC=B0*CHOURS**3-B1*CHOURS**2+B2*CHOURS-B3;
              DER.BO=CHOURS**3;
              DER.B1=-CHOURS**2;
              DER.B2=CHOURS:
              DER.B3=-1;
    OUTPUT OUT = B P = PREDB R=YRESID ;
GÓPTIONS DEVICE=TEK4014 VPOS 120;
    TITLE1 'UH-1 CLASS B (1977-1989) CUBIC EQUATION FIT';
     SYMBOL1 V=B ;
      SYMBOL2 V=DIAMOND .25 CM I=SPLINE;
PROC GPLOT;
   TI: CUBIC FIT UH-1 PRED. CLASS B INCIDENTS;
   PLOT PREDB * CHOURS = 2
        BINC * CHOURS = 1/OVERLAY;
  LABEL PREDB = 'CUM. CLASS B MISHAPS';
  LABEL HOURS = 'CUM. FLYING HOURS';
PROC PRINT UNIFORM;
   VAR HOURS CHOURS BINC PREDB ;
```

MORE...

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TOF:
    CHOURS=CUM FLIGHT HOURS (X10000);*/
•
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    AINC = CUM CLASS A INCIDENTS; */
/* BINC = CUM CLASS B INCIDENTS:*/
DATA UH1H:
    INPUT YEAR CHOURS AINC BINC:
    CARDS;
1974 6.42 12 3
1975 13.0 38 8
1976 19.8 55 10
1977 27.0 66 13
1978 34.0 88 16
1979 41.0 98 19
1980 48.7 110 19
1981 56.1 124 23
1982 62.8 141 32
1983 69.7 148 37
1984 75.9 162 39
1985 82.1 172 41
1986 88.6 181 47
1987 95.0 188 49
1988 - 3 197 49
                                                             MORE...
   10 CLE UH-1H CUM ACTUAL & PREDICTED CLASS B INCIDENTS 1974-1987;
      SAS PROGRAM TO FIT A CUBIC EQUATION TO CLASS B INCIDENTS
        FOR THE UH-1H AIRCRAFT. 1989 LINEAR FIT WAS POOR.
PROC NLIN;
     PARMS B0=0 B1=0 B2=0 B3=0;
           MODEL BINC=B0*CHOURS**3-B1*CHOURS**2+B2*CHOURS-B3;
              DER.BO=CHOURS**3;
              DER.B1=-CHOURS**2;
            " DER.B2=CHOURS;
             DER.B3=-1;
    OUTPUT OUT = B P = PRLIB F=:RESID ;
GOPTIONS DEVICE=TEK4014 VPOS 120;
    TITLE1 'UH-1H CLASS B (1977-1989) CUBIC EQUATION FIT';
     SYMBOL1 V=B ;
      SYMBOL2 V=DIAMOND .25 CM I=SPLINE;
PROC GPLOT:
   TITLE CUBIC FIT UH-1H CUM. CLASS B INCIDENTS;
   PLOT PREDB * CHOURS = 2
        BINC * CHOURS = 1/OVERLAY;
  LABEL PREDE = 'CUM. CLASS B MISHAPS';
  L REL 1-01FE = 'CUM' FLYING HOURS';
                                                             MORE...
LEGEND:
PROC PRINT UNIFORM;
   VAR CHOURS BINC PREDB ;
EOF:
```

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                                                               MORE...
TOF:
× 44
     CHOURS=CUM FLIGHT HOURS (X10000);*/
     AINC = CUM CLASS A INCIDENTS; */
/*
     BINC = CUM CLASS B INCIDENTS; */
DATA OH58;
    INPUT CHOURS AINC BINC:
     CARDS:
3.13 9 0
6.34 15 0
7,27 27 0
12.3 33 0
15.3 40 0
18.1 50 1
20.9 61 1
24.0 74 1
26.8 82 1
29.6 98 2
32.3 105 2
35.0 117 2
 , TITLE OH-58 CUM ACTUAL & PREDICTED CLASS B INCIDENTS 1974-1987;
      SAS PROGRAM TO FIT A CUBIC EQUATION TO CLASS B INCIDENTS
                                                               MORE...
./<sub>≱</sub>}
        FOR THE OH-58 AIRCRAFT. 1989 LINEAR FIT WAS POOR.
                                                                   ¥:/
PROC NLIN:
     PARMS B0=0 B1=0 B2=0 B3=0;
           MODEL BINC=B0*CHOURS**3-B1*CHOURS**2+B2*CHOURS-B3;
              DER.BO=CHOURS**3;
              DER.B1=-CHOURS**2:
              DER.B2=CHOURS;
              DER.B3=-1:
    OUTPUT OUT = B P = PREDB R=YRESID ;
GOPTIONS DEVICE=TEK4014 VPOS 120;
    TITLE1 'OH-58 CLASS B CUBIC EQUATION FIT';
     SYMBOL1 V=B :
      SYMBOL2 V=DIAMOND .25 CM I=SPLINE;
PROC GPLOT:
   TITLE CUBIC FIT OH-58 CUM. CLASS B INCIDENTS;
   PLOT PREDB * CHOURS = 2
        BINC * CHOURS = 1/OVERLAY;
  LABEL PREDB = 'CUM. CLASS B MISHAPS';
  LABEL CHOURS = 'CUM. FLYING HOURS';
 EGEND:
PROC PRINT UNIFORM:
   VAR CHOURS BINC PREDB ;
```

MORE...

```
TOF:
/*
     CHOURS=CUM FLIGHT HOURS (X10000);*/
/*·
     AINC = CUM CLASS A INCIDENTS:*/
/*
    BINC = CUM CLASS B INCIDENTS; */
DATA OH6:
    INPUT CHOURS AINC BINC:
     CARDS:
.302 4 0
.636 5 0
.992 5 0
1.32 8 0
1.61 8 0
1.94 8 0
2,27 9 0
2.63 10 1
2.98 11 1
3.25 11 1
3.50 12 1
3.75 12 1
4.06 13 1
   TITLE OH-6 CUM ACTUAL & PREDICTED CLASS B INCIDENTS 1974-1987:
                                                             MORE...
10
       SAS PROGRAM TO FI - 1 1 ZOUATION TO CLASS B INCIDENTS
       FOR THE OH-6 AIRCRAFT. 1989 LINEAR FIT WAS POOR.
÷÷
PROC NLIN:
    PARMS B0=0 B1=0 B2=0 B3=0;
           MODEL BINC=B0*CHOURS**3-B1*CHOURS**2+B2*CHOURS-B3;
              DER.BO=CHOURS**3;
              DER_B1=-CHOURS**2;
              DER.B2=CHOURS;
              DER.B3=-1;
   OUTPUT OUT = B P = PREDB R=YRESID ;
GOPTIONS DEVICE=TEK4014 VPOS 120;
    TITLE1 'OH-6 CLASS B CUBIC EQUATION FIT';
     SYMBOL1 V=B ;
      SYMBOL2 V=DIAMOND .25 CM I=SPLINE;
PROC GPLOT:
   TITLE CUBIC FIT OH-6 CUM. CLASS B INCIDENTS;
   PLOT PREDB * CHOURS = 2
        BINC * CHOURS = 1/OVERLAY;
 LABEL PREDB = 'CUM. CLASS B MISHAPS';
 LABEL CHOURS = 'CUM. FLYING HOURS';
LEGEND;
PROC PRINT UNIFORM;
                                                             MORE...
   VAR CHOURS BINC PREDB ;
EOF:
```

```
TOF:
./₩
     CHOURS=CUM FLIGHT HOURS (X 100,000);*/
/★
     AINC = CUM CLASS A INCIDENTS; */
    BINC = CUM CLASS B INCIDENTS; */
.√*
DATA TH55A;
    INPUT CHOURS AINC BINC;
     CARDS;
1,19 17 0
1.94 23 0
2.77 26 0
3.42 30 0
3.96 31 0
4.64 31 0
5.65 31 0
5.64 31 0
7.67 31 0
8.65 31 0
9.57 32 0
10.6 32 0
   TITLE TH-55 CUM ACTUAL & PREDICTED CLASS B INCIDENTS 1974-1987;
       SAS PROGRAM TO FIT A CUBIC EQUATION TO CLASS A INCIDENTS
                                                             MORE...
· •
        FOR THE TH-55 AIRCRAFT. 1989 LINEAR FIT WAS POOR.
                                                                 */
PROC NLIN;
     PARMS B0=0 B1=0 B2=0 B3=0;
           MODEL AINC=B0*CHOURS**3-B1*CHOURS**2+B2*CHOURS-B3;
              DER.BO=CHOURS**3;
              DER.B1=-CHOURS**2;
              DER.B2=CHOURS;
              DER.B3=-1;
    OUTPUT OUT = B P = PREDA R=YRESID ;
JOPTIONS DEVICE=TEK4014 VPOS 120;
    TITLE1 'TH-55 CLASS A CUBIC EQUATION FIT';
     SYMBOL1 V=B ;
      SYMBOL2 V=DIAMOND .25 CM I=SPLINE;
PROC GPLOT;
   TITLE CUBIC FIT TH-55 CUM. CLASS A INCIDENTS;
   PLOT PREDA * CHOURS = 2
        AINC * CHOURS = 1/OVERLAY;
  LABEL PREDA = 'CUM. CLASS A MISHAPS';
  LABEL CHOURS = 'CUM. FLYING HOURS':
LEGEND:
PROC PRINT UNIFORM;
   VAR CHOURS AINC PREDA ;
                                                              MORE...
```